



# **ASTER**

## **User's Guide**

**Part II**

**Level 1 Data Products**

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**ERSDAC**  
**Earth Remote Sensing Data**  
**Analysis Center**

# ASTER User's Guide

## Part II

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# 1. Level-1 Processing and Data Products Overview

*Introduction:* ASTER (Advanced Spaceborne Thermal Emission and Reflection Radiometer) is an advanced multispectral sensor that is a facility instrument selected by NASA to fly on the Terra polar orbiting spacecraft in December 1999, and covers a wide spectral region from visible to thermal infrared with 14 spectral bands with high spatial, spectral and radiometric resolution. The spectral bandpasses are shown in Table 2-1. The wide spectral region is covered by three telescopes, three VNIR (Visible and Near Infrared Radiometer) bands with a spatial resolution of 15 m, six SWIR (Short Wave Infrared Radiometer) bands with a spatial resolution of 30 m and five TIR (Thermal Infrared Radiometer) bands with a spatial resolution of 90 m. In addition one more telescope is used to see backward in the near infrared spectral band (band 3B) for stereoscopic capability that will produce a base-to-height ratio of 0.6. Please refer to Volume I for more details on the ASTER instrument and the science objective

This multi-telescope configuration necessitates inter-telescope band-to-band registration with an image matching technique in the Level-1 algorithm. The intra-telescope band-to-band registration for the SWIR bands is also carried out with the image matching technique to remove the parallax error due to the detector distribution on the focal plane.

The ASTER instrument has two types of Level-1 data, Level-1A and Level-1B data. Level-1A data are formally defined as reconstructed, unprocessed instrument data at full resolution. According to this definition the ASTER Level-1A data consist of the image data, the radiometric coefficients, the geometric coefficients and other auxiliary data without applying the coefficients to the image data to maintain the original data values. The Level-1B data are generated by applying these coefficients for radiometric calibration and geometric resampling.

All acquired image data are required to be produced to Level-1A. The ASTER Ground Data System (ASTER GDS) must handle a large amount of data, because the ASTER instrument has a high spatial resolution. The average data rate allocated to the ASTER instrument is limited to 8.3 Mbps which roughly corresponds to a duty cycle of 8 %. Therefore, the maximum daily data volume which the ASTER GDS must handle is 780 sets of 60 km x 60 km scenes and corresponds to about 80 GB daily. A maximum of 310 scenes per day are to be processed to Level-1B data in response to requests from users.

Table 2-1 Spectral passbands

Subsystem	Band No.	Spectral Range ( $\mu\text{m}$ )	Spatial Resolution
VNIR	1	0.52 - 0.60	15 m
	2	0.63 - 0.69	
	3N	0.78 - 0.86	
	3B	0.78 - 0.86	
SWIR	4	1.600 - 1.700	30 m
	5	2.145 - 2.185	
	6	2.185 - 2.225	
	7	2.235 - 2.285	
	8	2.295 - 2.365	
	9	2.360 - 2.430	
	10	8.125 - 8.475	
TIR	11	8.475 - 8.825	90 m
	12	8.925 - 9.275	
	13	10.25 - 10.95	
	14	10.95 - 11.65	

*End-to-End Processing Concept:* Figure 1-1 summarizes the end-to-end Level-1 data processing flow. Level-0 data, which are packetized in CCSDS (Consultative Committee for Space Data System) format and sent from EDOS (EOS Data and Operations System), are processed to the Level-0A data in the front-end processing module. The front-end processing module includes a depacketizing function to recover the instrument source data and then a demultiplexing function to separate image data into spectral bands with BSQ (Band Sequential) format. The SWIR and the TIR image data are realigned to compensate for a staggered configuration. Level-0A data consist of three groups, the VNIR data group, the SWIR data group and the TIR data group. Each group consists of the image data, the instrument supplementary data and the spacecraft ancillary data. The short term calibration data are included only for the TIR data group. In this stage the image data are not divided into scenes but kept in a continuous observation unit, that is, a long strip for more flexible scene selection.

The front-end processing is followed by the geometric system correction and generation of the radiometric coefficients. The system correction function mainly consists of the coordinate transformation of the line of sight vector by using the ancillary information from the spacecraft and the supplementary information from the instrument to identify the observing point in latitude/longitude coordinates on the earth's surface defined by the earth model (WGS-84). The radiometric coefficients specific for the strip unit are generated using real temperature values in the instrument supplementary data.

Level-0B data consist of the Level-0A data, the geometric system correction data and the radiometric correction data. Level-0B data products are used for the image matching and the cloud coverage calculation.

The SWIR parallax error is caused by the offset in detector alignment in the along-track direction (Figure 8a) and depends on the distance between the spacecraft and the observed earth's surface. For SWIR bands the parallax error corrections are carried out with the image matching technique or the coarse DEM data base depending on cloud coverage, by using Level-0B data.

For SWIR and TIR bands, the inter-telescope registration correction is carried out with VNIR band 2. The correction coefficient is evaluated by image matching between bands 2 and 6 for SWIR bands and between bands 2 and 11 for TIR bands.

The scene cutting is carried out on Level-0B data, according to the predetermined World Reference System (WRS). Each group of data are divided into scenes of 60 km in the along-track direction but includes 3 more km of data to provide an overlap of 5 % with neighboring scenes except for backward stereo band 3B. For band 3B the scene size is 81 km, including an additional overlap of 6 km to compensate for the terrain error contribution and a scene rotation for a large cross-track pointing

All geometric correction processes and scene cutting process are followed by a set of geolocation data generated for each scene. All geometric information is consolidated into a set of geolocation data expressed in latitude/longitude for every block.

The Level-1A data product consists of the image data, the radiometric coefficients, the geolocation data and the auxiliary data. The Level-1B data product can be generated by applying these data for radiometric calibration and geometric resampling.

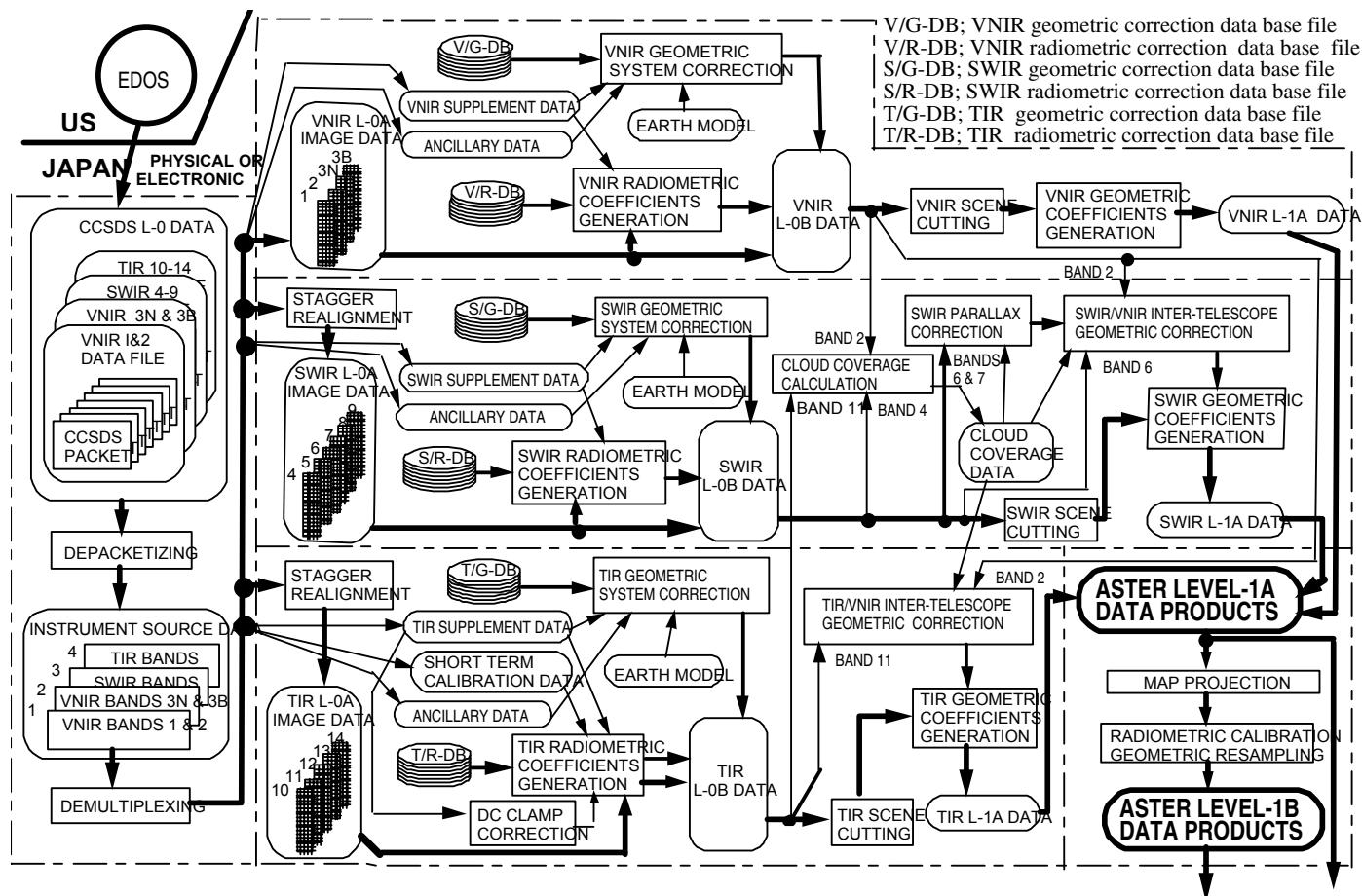


Figure 1-1 Summarized end-to-end processing flow

## 2. Level-1A Processing Algorithm

### 2.1. Level-0 Data

**Level-0 Production Data Set:** The Level-0 data which are sent from EDOS are packetized in the CCSDS format as shown in Figure 2-1. These packets are classified into four groups of data according to an APID (Application Process Identification) in the primary header of each packet. The group 1 data contain the data for VNIR bands 1 and 2. The group 2 data contain the data for VNIR bands 3N and 3B. The group 3 data contain the data for all SWIR bands. The group 4 data contain the data for all TIR bands. Each data group include science image data, instrument supplement data and spacecraft ancillary data.

ASTER is allocated the 64 APIDs that lie within the decimal equivalent range of 256 - 319. Different APIDs are allocated depending on data content (image data, instrument supplement data, or spacecraft ancillary data), data groups and operation modes (observation mode, calibration mode and test mode).

The packets are sorted out with the time tag data in the secondary header and then sorted out in the order of the image data, the instrument supplement data and the spacecraft ancillary data with different APIDs. Some packets contain both the supplement data and the ancillary data in the same packet. In a group with the same APID the packets are sorted out in the order of sequential counter data in the primary header.

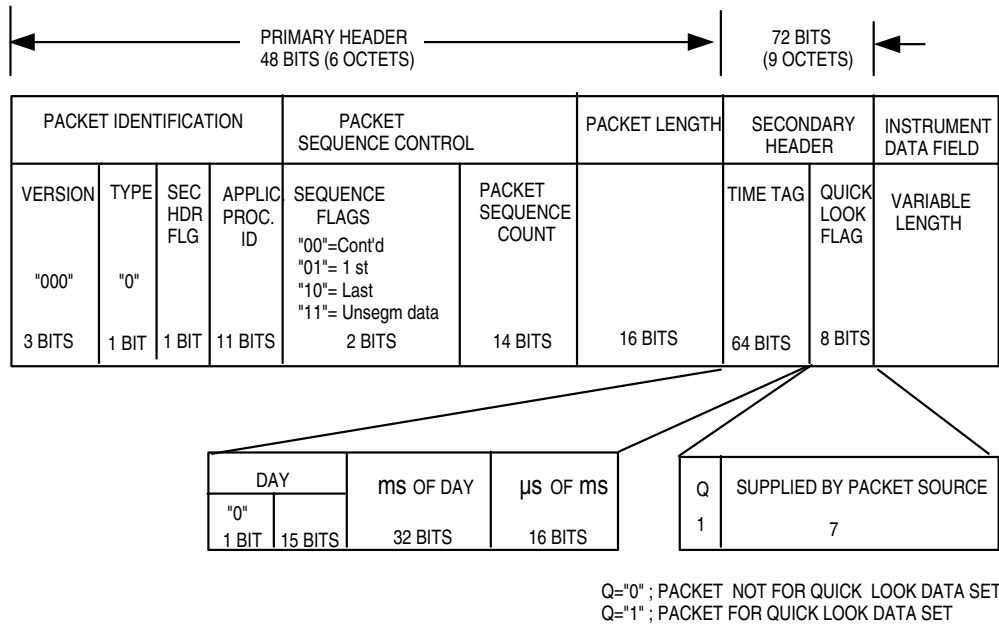


Figure 2-1 CCSDS Level-0 data packet format

## 2.2. Front-end Processing

*Depacketizing of CCSDS Level-0 Data:* The packets of each group are depacketized and aligned to recover the unpacktized instrument source data by using a sequential counter, flags in the primary header and time tags in the secondary header. In the instrument source data format, the spectral band information is multiplexed with the image in BIP (Band Interleaved by Pixel) format as shown in Figure 2-2. Each swath line of image data is appended by the instrument supplement data and spacecraft ancillary data specific for the swath line.

VNIR (1): Bands 1 & 2		
IMAGE DATA (BIP FORMAT) 65,600 BITS	VNIR SUPPLEMENT DATA 400 BITS	ANCILLARY DATA 512 BITS
VNIR (2): Bands 3N & 3B		
IMAGE DATA (BIP FOMAT) 65,600 BITS	VNIR SUPPLEMENT DATA 400 BITS	ANCILLARY DATA 512 BITS
SWIR: Bands 4 - 9		
IMAGE DATA (BIP FORMAT) 98,304 BITS	SWIR SUPPLEMENT DATA 328 BITS	ANCILLARY DATA 512 BITS
TIR: Bands 10 - 14		
IMAGE DATA (BIP FORMAT) 453,120 BITS	TIR SUPPLEMENT DATA 79,312 BITS	ANCILLARY DATA 512 BITS

Figure 2-2 Instrument source data format

*Demultiplexing Instrument Source Data:* The instrument source data are demultiplexed to separate image data for every spectral band into BSQ (Band Sequential) format. Here, we have the data rearranged in three groups, that is, the VNIR data group, the SWIR data group and the TIR data group. Each data group consists of the image data for each spectral band, the supplement data and the ancillary data. For only the TIR data group, the short term calibration obtained at the beginning of each observation is included. The supplementary data are necessary for all of the data groups to make it possible to process them independently. In this stage the image data are not divided into scenes but kept in one continuous observation unit, that is, a long strip of image data for more flexible scene selection. This data set is defined as Level-0A data which is a tentative product only used during processing.



*Image Data Stagger Realignment:* For SWIR and TIR image data the Level-0 pixel addresses are changed, so that all pixels for each band lie on one line, compensating for the staggered configuration described below. This realignment process is carried out not only to have more exactly aligned image data without resampling for the image matching process but also to simplify subsequent processes.

The SWIR subsystem uses electronically scanned linear detector arrays for each band to obtain one data line simultaneously in the cross-track direction for each scan period. These detector arrays are separated for odd and even numbered detectors with a staggered configuration as shown in Figure 2-3(a). The realignment for SWIR pixel addresses is carried out to compensate for the difference from the center line between the odd and the even lines. The stagger offset value to the center line can be set to  $\pm 1$  pixel with a good approximation.

TIR images are obtained by mechanical scanning with 10 detectors for each spectral band, that is, 50 detectors in total. Ten detectors for each band are arranged with the staggered configuration in the cross-track direction as shown in Figure 2-3(b). The realignment for TIR pixel addresses is carried out to compensate for the difference from the center line between the odd and the even lines. The stagger offset value to the center line can be set to  $\pm 4$  pixels with a good approximation.

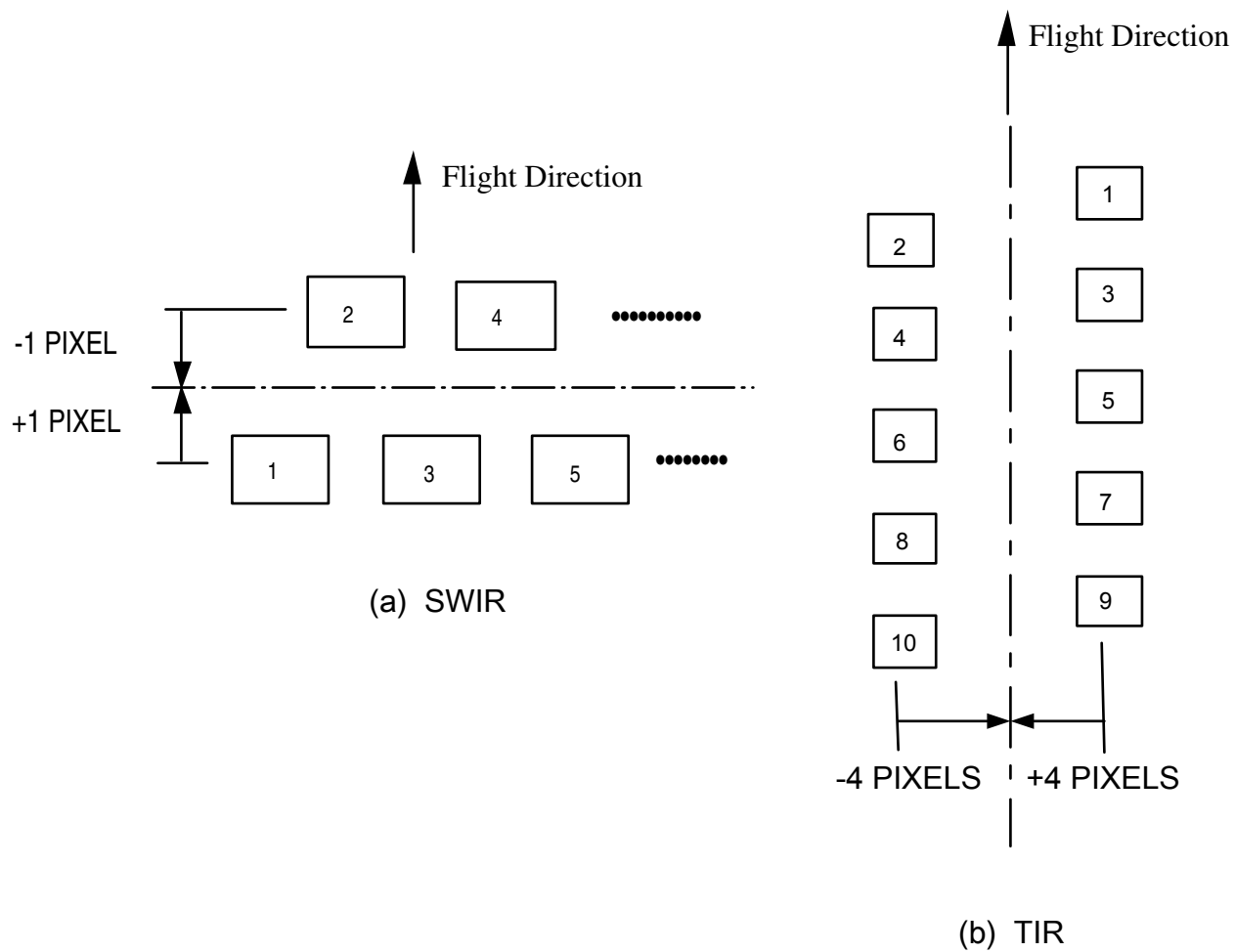


Figure 2-3 Realignment positions of staggered SWIR and TIR pixels

## 2.3. Radiometric Correction

*Common:* Radiometric coefficients are generated in two steps. The first step is an off-line process from Level-1 data product generation to prepare radiometric coefficients at predefined reference temperature. These coefficients will be effective for a long time, depending on the instrument stability, and available in radiometric correction data base files along with the temperature coefficients. One set of offset and sensitivity data are necessary for the VNIR and SWIR bands and one set of offset, linear sensitivity and nonlinear sensitivity data are necessary for the TIR bands. Destriping parameters will be generated from the image data if necessary, analyzing the image data obtained during the initial checkout operation period.

The second step is an on-line process executed during Level-1 product generation to correct the radiometric coefficients for instrument conditions such as detector temperature and dewar temperature, which may change for every observation.

Destriping parameters will be generated from the image data if necessary, analyzing the image data obtained during the initial checkout operation period.

*Generation of Radiometric Correction Data Base:* The radiometric coefficients for the reference temperatures were evaluated during the preflight test period using integration spheres, followed by a regular evaluation during the inflight period with on-board calibration and vicarious calibration data, and then updated if necessary. On-board calibration is scheduled every 17 days to check the stability.

These radiometric coefficients are available in the radiometric correction data base with temperature coefficients as on-line parameter files applied in the Level-1 processing. All temperature coefficients were prepared during the preflight test period and will be used throughout the mission period.

The required absolute accuracies are 4% ( $\sigma$ ) for VNIR and SWIR, and 1 to 3K for TIR depending on target temperatures.

*Generation of VNIR Observation-Unit-Specific Radiometric Coefficients:* Figure 2-4(a) shows the VNIR radiometric coefficients generation algorithm flow. Detector temperature is the only reference parameter by which the radiometric coefficients (offset and sensitivity) have to be corrected for a specific observation unit data set. Under normal operating conditions within the designed temperature range, this correction process will not be necessary, since the correction value is expected to be very small

*Generation of SWIR Observation-Unit-Specific Radiometric Coefficients:* Figure 2-4(b) shows the SWIR radiometric coefficients generation algorithm flow. Detector temperature and dewar temperature are the reference parameters by which the radiometric coefficients (offset and sensitivity) have to be corrected for a specific observation unit data. Under normal operating conditions the detector temperature is controlled to within  $\pm 0.2$  K at around 77 K. Therefore the correction process for detector temperature will not be necessary as long as the SWIR is operating normally .

The dewar temperature correction, which is necessary for compensating only for thermal radiation from the dewar, is applied only to the offset, since the SWIR detector is sensitive to room temperature thermal radiation up to 5  $\mu\text{m}$  and the band pass filter can not completely remove this out-of-band radiation. The dewar temperature is used as a representative value for the internal thermal radiation.

*Generation of TIR Observation-Unit-Specific Radiometric Coefficients:* Figure 2-4(c) shows the TIR radiometric coefficients generation algorithm flow. The linear and the non-linear sensitivities coefficients in the data base are corrected only by the detector temperature. Under normal operating conditions the detector temperature is within  $\pm 0.2$  K at around 80 K. Therefore, correction for the detector temperature will not be necessary as long as TIR is operating normally.

The offset data, common throughout the observation unit, are generated from the short term calibration data acquired at the beginning of each observation by using the blackbody temperature in the TIR supplementary data.

The correction for the offset data due to chopper temperature is calculated using the chopper temperature changes from the short term calibration period. The chopper temperature correction is carried out on the TIR Level-0A image data with the DC clamp correction. This correction will be possible for each scan data (each ten lines of data in the along-track direction), since the chopper temperature data is included in the supplementary data.

This image data correction will result in slightly different TIR Level-0B image data DN values from Level-0 data. The Level-0 data is digitized to 12 bits. The LSB (Least Significant Bit) value of Level-0B image data is very small compared to  $NE\Delta T$  (less than one third for the 300 K target). Therefore, this difference will not give rise to any significant round-off error.

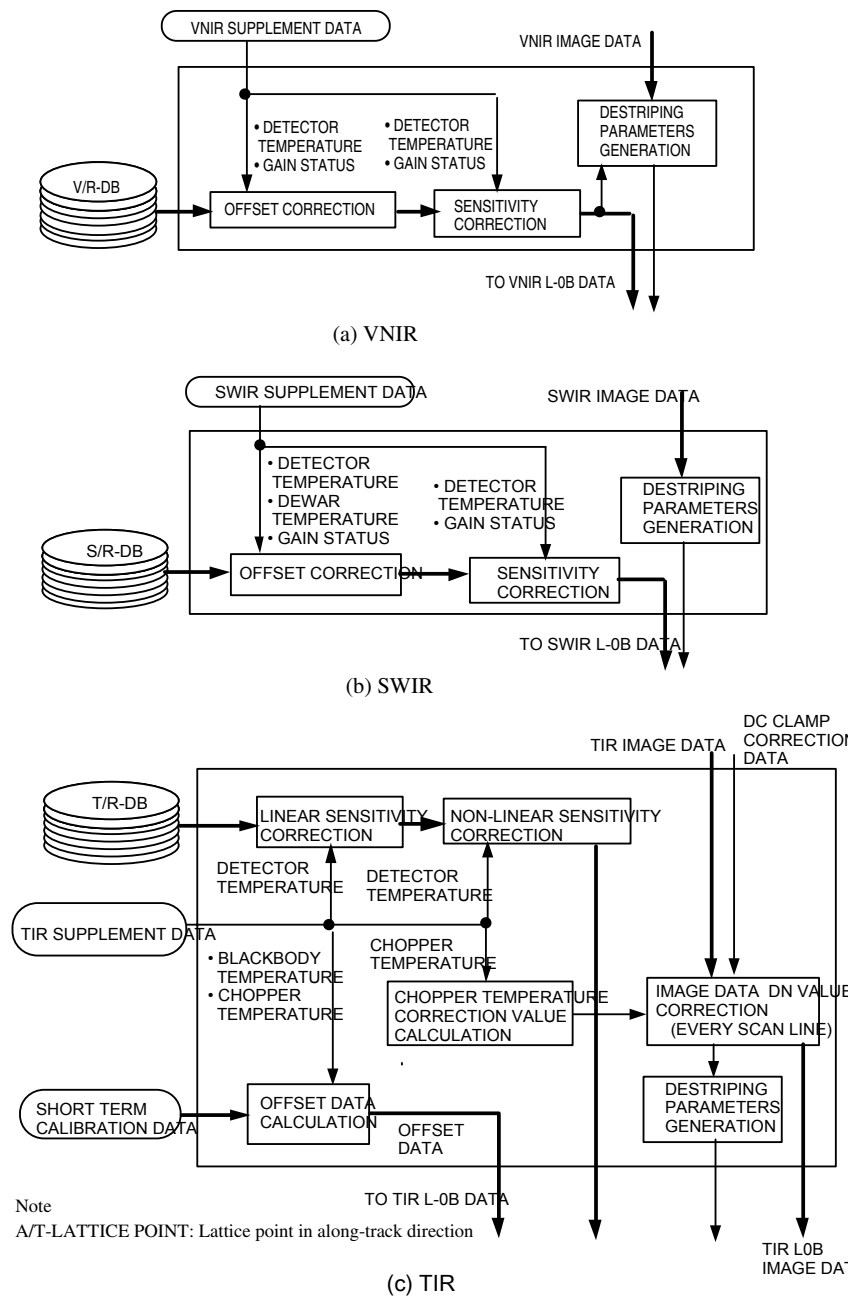


Figure 2-4 Radiometric correction coefficients generation flow

**TIR DC Clamp Correction:** TIR output voltage is clamped at  $-1.4 \text{ V} \pm \bullet V_n$  for bands 10-12 and  $-0.9 \text{ V} \pm \bullet V_n$  for bands 13 and 14 when the chopper plate is observed by the detectors at every scan. The small voltage  $\bullet V_n$  is the noise voltage at the moment of the clamp which changes randomly in every scan and must be corrected to exactly set the clamp voltage.

Figure 2-5 shows the TIR DC clamp correction flow. The exact clamped output (DN value) is available in the TIR supplementary data as the chopper data. The chopper data in the one previous scan are used for the correction. The 100 chopper data acquired for one scan are averaged to reduce noise component, followed by the DC clamp error calculation which corresponds to  $\pm \bullet V_n$ . This clamp error is transferred to the TIR radiometric correction module to subtract it from the TIR Level-0A image data to generate Level-0B image data.

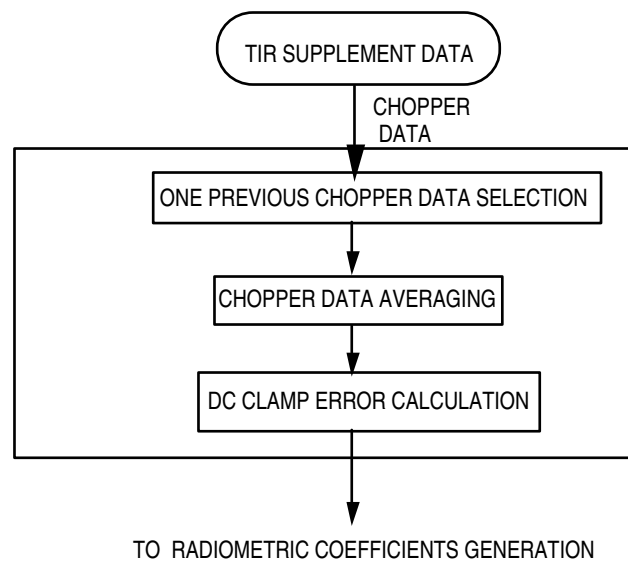


Figure 2-5 TIR DC clamp correction

## 2.4. Geometric Correction Scheme

*Common:* The geometric correction process consists of three steps. The first step is an off-line process from Level-1 data product generation to prepare geometric parameters such as the line-of-sight vectors of detectors and the pointing axes information. These parameters are effective for a long time, depending on the instrument stability, and are recorded in the geometric correction data base files.

The second step is an on-line process executed during Level-1 product generation to identify the observing points of detectors, using the geometric parameters available in the geometric correction data base files and the engineering information supplied by the instrument and the spacecraft. This process is called “Geometric System Correction”.

The third step is also an on-line process to enhance band-to-band registration accuracy by image matching techniques. This step consists of two parts. One is the SWIR parallax correction process. Another is the inter-telescope registration process.

*Generation of Geometric Correction Data Base:* The geometric correction data base files contain the line-of-sight vectors of the detectors, the pointing axis vectors and the conversion coefficients from encoder values to angles. The line-of-sight vectors of the detectors and the pointing axis vectors were evaluated toward the Navigation Base Reference of the spacecraft during the preflight test period using the collimator data and the alignment data on the spacecraft.

These geometric parameters will be corrected after launch through the validation activity using GCPs (Ground Control Points) and band-to-band image matching techniques.

## 2.5. Geometric System Correction

*General Description* : The geometric system correction is the rotation and the coordinate transformation of the line of sight vectors of detectors to the earth Greenwich coordinate system using only the engineering information from the instrument and the spacecraft to identify the observed points by the detectors. The observed point on the surface is identified by the intersection of the earth's surface and an extended line-of-sight vector. The engineering information from the instrument and the spacecraft are called the supplementary data and the ancillary data, respectively. The geometric system correction is almost the same for the three subsystems, except for selected numbers of vectors to be transformed. Figure 2-6 shows the geometric system correction flow.

The image data are divided into blocks for both the cross-track and the along-track directions. The block sizes are as follows.

VNIR bands 1, 2, 3N:	410 x 400 pixels
VNIR band 3B:	500 x 400 pixels
SWIR all bands:	20 x 20 pixels
TIR all bands:	72 x 70 pixels

These values were decided by considering the distortion of optical images on the focal plane in the cross-track direction and spacecraft stability in the along-track direction. The coordinate transformations are carried out only for the line-of-sight vectors of selected detectors. The numbers of the selected detectors are 11, 104 and 11 for VNIR, SWIR and TIR bands, respectively, which correspond to the number of the corner for each block of Level-0 images in the cross-track direction. Dummy detectors will have to be introduced to compensate for and then to completely define the block at the end of the cross-track direction.

The geometric system correction is divided into several parts as follows:

- (1) The pointing correction
- (2) The coordinate transformation from Navigation Base Reference of the spacecraft to the Orbital Reference Frame
- (3) The coordinate transformation from the Orbital Reference Coordinate Frame to the Earth Inertial coordinate Frame
- (4) The coordinate transformation from the Earth Inertial Coordinate Frame to the Earth Greenwich Coordinate Frame
- (5) Identification of the intersection of the Earth surface and an extension of the line of sight vector



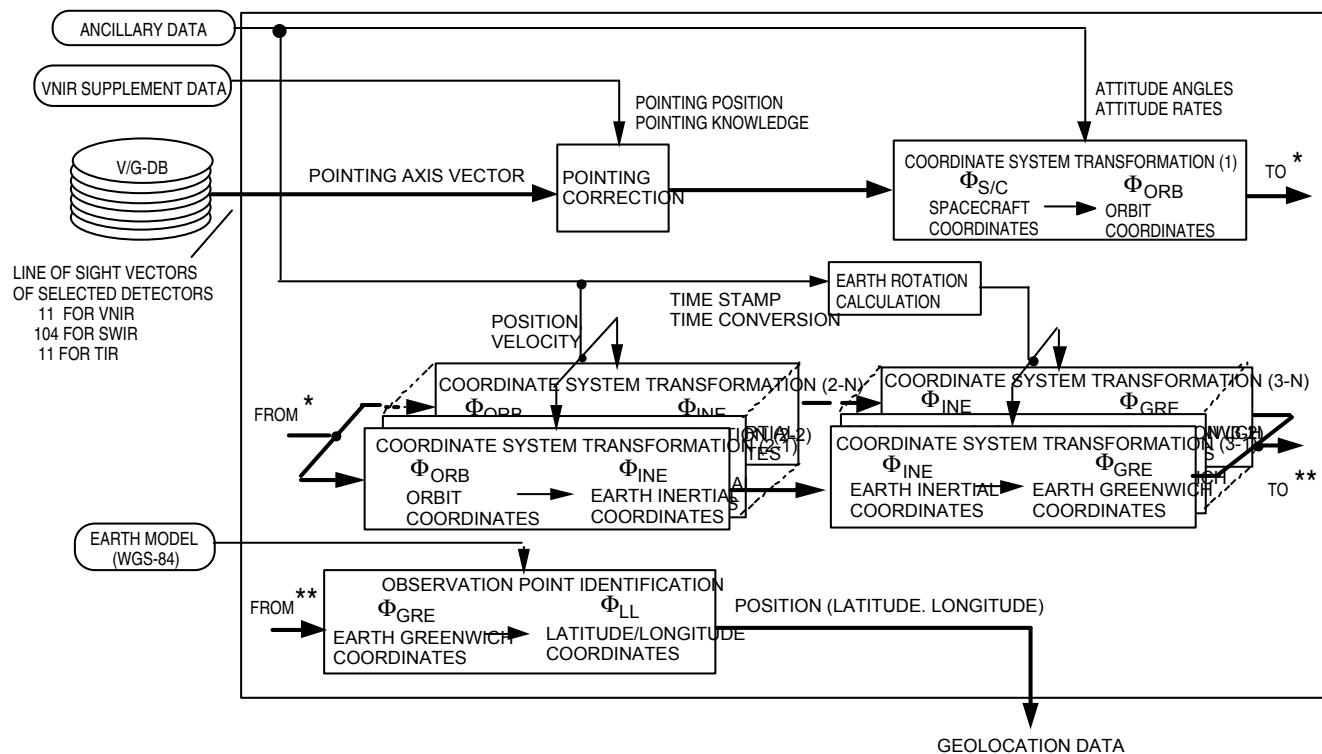


Figure 2-6 Geometric system correction flow

**Pointing Correction-** : The line of sight vectors in the geometric data base are those for the reference pointing angles (nominal nadir direction). The line of sight vectors are changed using the pointing position and knowledge from the supplementary data. The pointing axes information in the geometric data base are used for the transformation of the line-of-sight vectors due to change in the pointing position.

The line of sight vector changes with the rotation for the pointing axis by an angle of  $\beta$  from  $S_0$  to  $S$  as follows.

$$\begin{bmatrix} S_X \\ S_Y \\ S_Z \end{bmatrix} = M^{-1} \begin{bmatrix} 1 & 0 & 0 \\ 0 & \cos \beta & -\sin \beta \\ 0 & \sin \beta & \cos \beta \end{bmatrix} M \begin{bmatrix} S_{0X} \\ S_{0Y} \\ S_{0Z} \end{bmatrix} \quad (2-1)$$

where

$S_{0x}, S_{0y}, S_{0z}$ : x, y, z components of the line of sight vector  $S_0$  before pointing,  
 $S_x, S_y, S_z$ : x, y, x components of the line of sight vector  $S$  after pointing,

$$M \equiv \begin{bmatrix} \cos \theta_{yaw} & \sin \theta_{yaw} & 0 \\ -\sin \theta_{yaw} & \cos \theta_{yaw} & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} \cos \theta_{pitch} & 0 & -\sin \theta_{pitch} \\ 0 & 1 & 0 \\ \sin \theta_{pitch} & 0 & \cos \theta_{pitch} \end{bmatrix}, \quad (2-2)$$

$$M^{-1} \equiv \begin{bmatrix} \cos \theta_{pitch} & 0 & \sin \theta_{pitch} \\ 0 & 1 & 0 \\ -\sin \theta_{pitch} & 0 & \cos \theta_{pitch} \end{bmatrix} \begin{bmatrix} \cos \theta_{yaw} & -\sin \theta_{yaw} & 0 \\ \sin \theta_{yaw} & \cos \theta_{yaw} & 0 \\ 0 & 0 & 1 \end{bmatrix}, \quad (2-3)$$

$$\Delta \theta_{yaw} \equiv \sin^{-1}(P_y), \quad (2-4)$$

$$\Delta \theta_{pitch} \equiv -\tan^{-1}(P_z/P_x), \quad (2-5)$$

$P_x, P_y, P_z$ : x, y, z componets of the pointing axes unit vector in the NBR Coordinate Frame.

Figure 2-7 shows the relation between the pointing axis and the NBR Coordinate Frame. The angles  $\Delta \theta_{yaw}$  and  $\Delta \theta_{pitch}$  are the yaw and the pitch rotation angles, respectively, to coalign the  $X_{NBR}$  to the pointing axis.

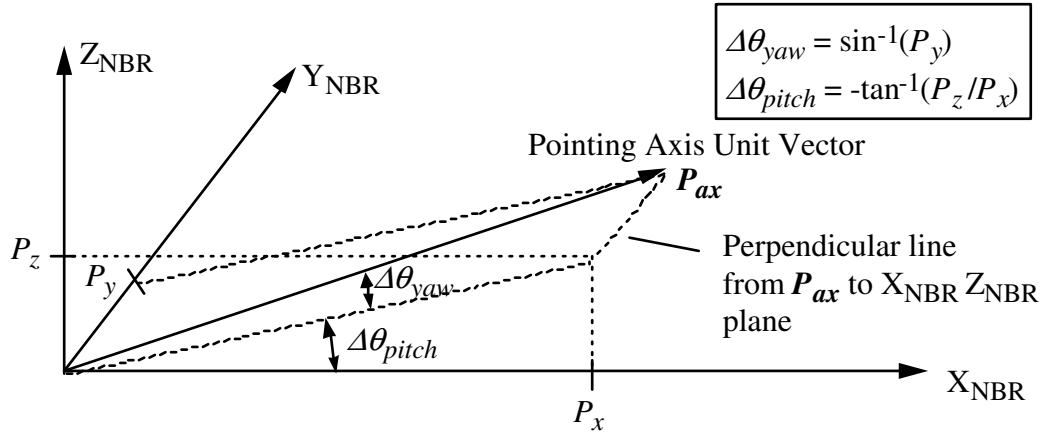


Figure 2-7 Pointing Axis Vector in NBR Coordinate Frame

*Spacecraft-to-Orbit Coordinates :* The spacecraft coordinates are slightly different from the orbit coordinates. The difference originates from the spacecraft attitude control accuracy and is provided as the attitude angle data in the spacecraft ancillary information. The orbit coordinate system is right-handed and orthogonal. The +z-axis is a line from the spacecraft center of mass to the center of the earth. The +y-axis is a line normal to the z-axis and the spacecraft instantaneous velocity vector (negative orbit normal direction). The x-axis completes the right hand set. This process is carried out by using attitude angles and rates in the ancillary data.

The line of sight vectors in the Spacecraft NBR Coordinate Frame can be converted to the expression in the Orbital Reference Frame using the attitude angle data in the spacecraft ancillary data as follows.

$$\mathbf{S}_{OR} = F_{SO\bullet yaw} F_{SO\bullet pitch} F_{SO\bullet roll} \mathbf{S} , \quad (2-6)$$

where  $\mathbf{S}$  : the line of sight vector expressed in the NBR Coordinate Frame,  
 $\mathbf{S}_{OR}$  : the line of sight vector expressed in the Orbit Reference Coordinate Frame,

$$F_{SO\bullet roll} \equiv \begin{bmatrix} 1 & 0 & 0 \\ 0 & \cos(-\alpha_{roll}) & \sin(-\alpha_{roll}) \\ 0 & -\sin(-\alpha_{roll}) & \cos(-\alpha_{roll}) \end{bmatrix} , \quad (2-7)$$

$$F_{SO\bullet pitch} \equiv \begin{bmatrix} \cos(-\alpha_{pitch}) & 0 & -\sin(-\alpha_{pitch}) \\ 0 & 1 & 0 \\ \sin(-\alpha_{pitch}) & 0 & \cos(-\alpha_{pitch}) \end{bmatrix} , \quad (2-8)$$

$$F_{SO \rightarrow yaw} \equiv \begin{bmatrix} \cos(-\alpha_{yaw}) & \sin(-\alpha_{yaw}) & 0 \\ -\sin(-\alpha_{yaw}) & \cos(-\alpha_{yaw}) & 0 \\ 0 & 0 & 1 \end{bmatrix}, \quad (2-9)$$

$\alpha_{roll}$ ,  $\alpha_{pitch}$ ,  $\alpha_{yaw}$  : roll, pitch, yaw components of the attitude data, respectively, in the spacecraft ancillary data

*Orbit-to-Earth Inertial Coordinates* : This process is the coordinate transformation to earth-centered coordinates in inertial space. Two-dimensional array vectors can be obtained by this transformation using the spacecraft movement. The array dimension for one observation depends on each observation period, that is, number of pixels in the along-track direction. This process is carried out by using position and velocity information in the ancillary data. The Precession and the Nutation effects are considered to be the more accurate geolocation data, since the spacecraft position information is based on the mean of the J2000.0 coordinate frame, which is the earth inertial coordinates at noon of January 1st, 2000. The line of sight vectors in the Orbital Reference Coordinate Frame can be converted to the expression in the Earth Inertial Coordinate Frame as follows.

$$S_{EI} = F_{OI} S_{OR}, \quad (2-10)$$

where  $S_{OR}$  : the line of sight vector expressed in the Orbit Reference Coordinate Frame,  
 $S_{EI}$  : the line of sight vector expressed in the Earth Inertial Coordinate Frame,

$$F_{OI} \equiv (T_x \ T_y \ T_z), \quad (2-11)$$

$T_x \ T_y \ T_z$  : unit vector components of x, y and z axes of the Orbital Coordinate Frame expressed in the Earth Inertial Coordinate Frame and defined as

$$\begin{aligned} T_x &\equiv T_y \times T_z \\ T_y &\equiv \text{unit}(-R \times V) \\ T_z &\equiv \text{unit}(-R) \end{aligned} \quad (2-12)$$

$R, V$  : the spacecraft position and velocity vectors expressed in the Earth Inertial Frame

For more accurate calculation the Precession matrix  $P$  and the Nutation matrix  $N$  shall be applied to the line of sight vector  $S_{EI}$  in the Earth Inertial Coordinate Frame.

*Earth Inertial-to-Earth Fixed coordinates* : This process is the coordinate transformation to the earth centered and earth-fixed coordinates, and carried out by using the earth rotation values calculated from the time information in the ancillary data. The UTC time, which is provided from the spacecraft, is converted to the UT1 to calculate the exact earth rotation angle.

The line of sight vectors in the Earth Inertial Coordinate Frame can be converted to the expression in the Earth Fixed Coordinate Frame as follows.

$$\mathbf{S}_{EF} = \mathbf{F}_{IF} \mathbf{S}_{EI}, \quad (2-13)$$

where  $\mathbf{S}_{EI}$  : the line of sight vector expressed in the Earth Inertial Coordinate Frame,  
 $\mathbf{S}_{EF}$  : the line of sight vector expressed in the Earth Fixed Coordinate Frame,

$$\mathbf{F}_{IF} \equiv \begin{bmatrix} \cos \theta_g & \sin \theta_g & 0 \\ -\sin \theta_g & \cos \theta_g & 0 \\ 0 & 0 & 1 \end{bmatrix}, \quad (2-14)$$

$\theta_g$  : Greenwich true sidereal hour angle.

*Earth Surface Identification* : The observation point is identified from the intersection of the earth surface and an extension of the line-of-sight vector. The WGS-84 is used as the earth surface model.

The observing earth surface can be identified calculating the crossing point between the extension line of the LOS vector and the earth surface. The extension line of the LOS vector can be expressed as follows.

$$\begin{aligned} x &= X + S_{EF \cdot x} r \\ y &= Y + S_{EF \cdot y} r \\ z &= Z + S_{EF \cdot z} r \end{aligned} \quad (2-15)$$

where  $S_{EF \cdot x}$ ,  $S_{EF \cdot y}$ ,  $S_{EF \cdot z}$  : x, y, z components of the LOS vector  $\mathbf{S}_{EF}$  in the Earth Fixed Coordinate Frame,

X, Y, Z : x, y, z components of the Spacecraft position vector in the Earth Fixed Coordinate Frame

r : parameter.

The earth surface can be expressed as follows.

$$(x^2 + y^2)/a^2 + z^2/b^2 = 1, \quad (2-16)$$

where

$$\begin{aligned} a &= 6378136 \text{m} \quad (\text{Earth radius at equator--WGS-84}), \\ b &= a(1 - f) \quad (\text{Earth radius at pole----WGS-84}), \\ f &= 1/298.2572. \end{aligned} \quad (2-17)$$

The intersection can be calculated from eqs.(2-15) and (2-16). When the observing point is expressed as  $P_x$ ,  $P_y$ , and  $P_z$ , the geocentric latitude  $\psi$  and the longitude  $\lambda$  can be expressed as follows.

$$\psi = \tan^{-1}\{P_z/(P_x^2 + P_y^2)^{1/2}\}, \quad (2-18)$$

$$\lambda = \tan^{-1}(P_y/P_x). \quad (2-19)$$

**System Correction Accuracy :** Table 2-2 shows the pixel geolocation knowledge as a result of the geometric system correction considering both the spacecraft and the instrument contributions. Total ASTER pixel geolocation knowledge is decided by the spacecraft position knowledge, the spacecraft pointing knowledge and ASTER pointing knowledge. Only the pixel geolocation knowledge of VNIR is considered, since the SWIR and TIR bands will be coregistered to VNIR band 2 as a reference band in the Level-1 processing.

Table 2-2 Pixel Geolocation Knowledge

		Specification	Dynamic Error (3 $\sigma$ )	Static Error (3 $\sigma$ )
Along-track (m)	Spacecraft* <sup>1</sup>	±342	±28	±111
	ASTER/VNIR	±205	±38	±99
	Total	±431* <sup>2</sup>	±47	±149
Cross-track (m)	Spacecraft* <sup>1</sup>	±342	±25	±148
	ASTER/VNIR	±205	±48	±103
	Total	±437* <sup>2</sup>	±54	±180

\*1: Three non-optimal 9 minute TDRS contacts per orbit, GJM2 Geopotential ( 30 x 30), solar flux of 175, 5 % Cd error, TDRS ephemeris error of 75 meters. Two star trackers, rigid body/low frequency pointing knowledge error removed

\*2: Slightly larger than RSS of two values (Spacecraft and ASTER instrument), because of unallocated margin.

The geometric system correction accuracy will be regularly checked through the geometric validation activity using GCPs during the normal operation period.

## 2.6. Parallax Correction

The large offset among SWIR bands in the along-track direction shown in Figure 2-88(a) gives rise to parallax errors for band-to-band registration depending on the distance between the instrument and the targeted ground. The algorithm used for SWIR intra-telescope registration is a combination of the image matching correlation method and the coarse DEM method. The ETOPO-030 prepared by EROS Data Center will be used as the coarse DEM.

A criterion of selecting band for image matching is good correlation, consequently spectrally close. Bands 6 and 7 are selected, although any other combination of neighbouring bands in the 2 $\mu$ m region will be available for this purpose. As shown in Figure 8(a), detector arrays of bands 6 and 7 are located at opposite ends of the focal plane to increase the parallax detection sensitivity.

Figure 2-8(b) shows the parallax correction flow. In order to eliminate this band-to-band misregistration due to the parallax, the SWIR parallax correction process handles the image data as follows.

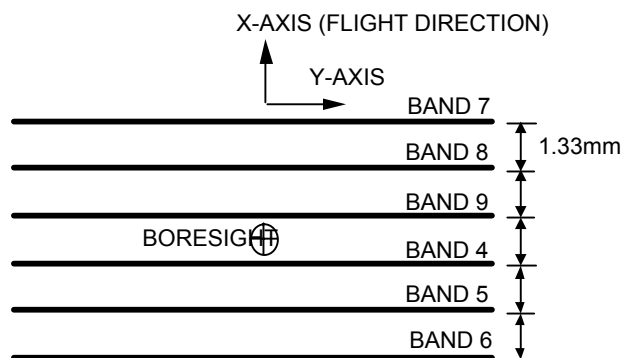
- (1) Select band 7 as a moving window image and band 6 as a target image. The moving window is selected such that its center corresponds to the lattice point at which the geometric system correction is carried out. The target image is selected to cover the search area. The moving window image (band 7) size is 21 by 21 pixel.
- (2) Select only cloud free windows for the correlation. Use the coarse DEM data for cloudy windows.
- (3) Carry out the radiometric correction for the two window images
- (4) Find correlation coefficients by moving the window in the along-track direction.
- (5) Find the highest correlation point in sub-pixel units by interpolating the correlation data calculated in pixel units.
- (6) Evaluate the image matching quality. Criteria for judgment are the correlation coefficient and the deviation from the predetermined value from the coarse DEM data. The threshold value for the correlation coefficient and the threshold deviation for the predetermined value are set to 0.7 and 20% deviation, respectively, in the initial stage. Use the parallax errors calculated by the coarse DEM if the image matching quality can not meet the criterion mentioned above.
- (7) Evaluate the parallax error of all SWIR bands for the nadir direction.

The registration error due to the parallax is evaluated at every lattice point (the corner point of the block) and expressed with the pixel unit for the Level-0B image data in the along-track direction.

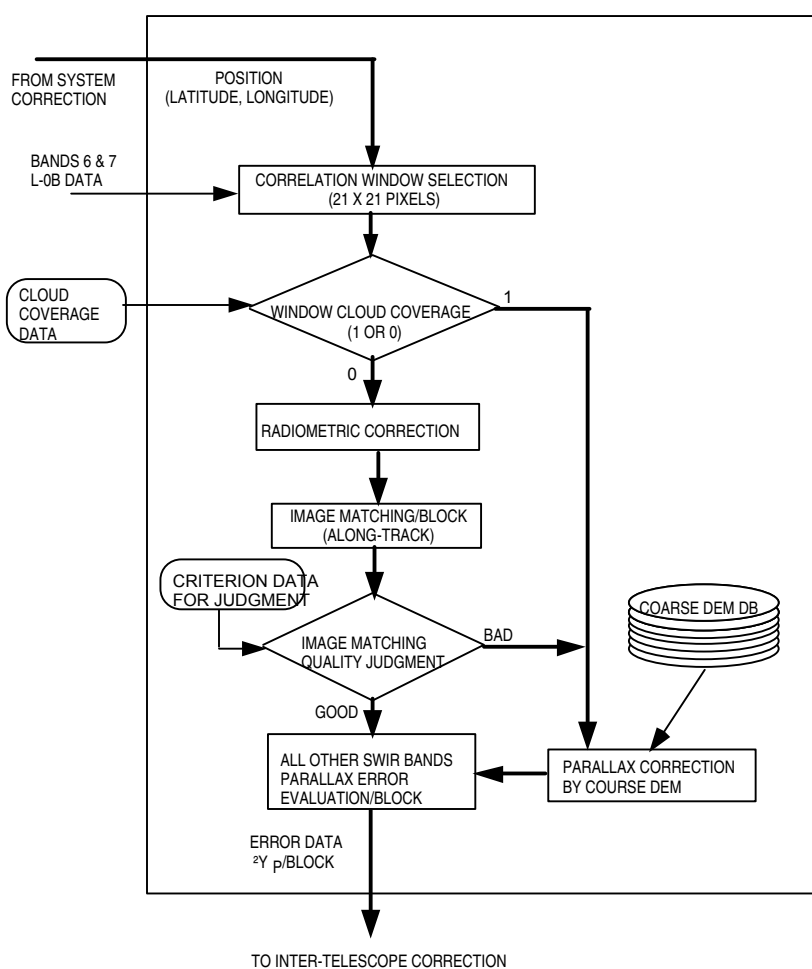
An evaluation of the algorithm was carried out using simulation images generated from airborne sensors and Landsat TM images. Miss-registration due to parallax error was intentionally introduced to

these images using DEM data. Judging from these results 99 % of the data are within 0.3 pixels for all images.





(a) SWIR focal plane configuration



(Note:  $^2Y_p$ : Parallax error in pixel unit for along-track direction)

(b) Parallax correction flow

Figure 2-8 SWIR focal plane configuration and parallax correction flow

## 2.7. Inter-telescope Band-to-band Registration Correction

The ASTER instrument configuration with the multi-telescopes necessitates routine processing for the inter-telescope band-to-band registration on the ground, unless the boresights are stable enough during the mission life to keep the initial state after launch. The pointing change mechanism is also a source of misregistration because of limited accuracy of pointing position knowledge.

The important feature to be stressed here for the inter-telescope registration is that a set of correction parameters will be valid and can be applied to all images as long as the pointing is kept at the same position and the elapsed time since a previous parameter setting is short enough for boresight stability. The inter-telescope registration correction will be carried out every observation unit, using reference bands of each telescope (band 2 for VNIR, band 6 for SWIR and band 11 for TIR).

There is no strong reason for this reference band selection, although a matching experiment was carried out using simulated image data. Band 2 was slightly better as a reference band than other VNIR bands because of a relatively large atmospheric absorption in band 1 and a peculiarity of band 3 for vegetation. Band 6 is selected as reference band of SWIR because it is one of the bands used for parallax correction. Band 11 is selected as reference band of TIR without any major reason.

Figure 2-9 shows the inter-telescope geometric correction flows for VNIR band 2. This correction process is carried out as follows.

- (1) Start the registration error calculation at the beginning of each observation unit.
- (2) Select band 2 as a moving window image and band 6 or 11 as a target image. The moving window is selected such that its center corresponds to the lattice point at which the geometric correction is carried out. The target image is selected to cover the search area. The moving window image size is 41 x 41 pixels. The target window size is larger than the moving window to cover the search area.
- (3) Select a cloud free window for the correlation by repeating the previous item (2).
- (4) Carry out the radiometric correction for the cloud free images
- (5) Find the correlation coefficients by moving the moving window in pixel units in both along-track and the cross-track directions.
- (6) Find the point of highest correlation in sub-pixel units by interpolating the correlation data calculation in the pixel unit.
- (7) Evaluate the image matching quality. The criteria for judgment is the correlation coefficient. The threshold value for the correlation coefficient is set to 0.7 in the initial stage.
- (8) Repeat the process from item (2) to item (7) a large number of times (between 100 and 200), selecting other target images, to enhance the accuracy by averaging.
- (9) If the preset number of error data can not be reached in an observation unit, the inter-telescope registration correction will not be carried out. The failed information will be reported.

- (10) Exclude the error data which deviates over  $3\sigma$  value from the average.
- (11) The obtained number of the effective error data is averaged to generate a set of final error data.
- (12) Calculate  $3\sigma$  value to evaluate the accuracy.

An evaluation of the algorithm was carried out using simulated images generated from Landsat TM and airborne sensor images. The registration accuracies in  $3\sigma$  for SWIR/VNIR were 0.054 SWIR pixels and 0.051 SWIR pixels in the cross-track and the along-track directions, respectively. The registration accuracies in  $3\sigma$  for TIR/VNIR were 0.050 TIR pixels and 0.044 TIR pixels in the cross-track and the along-track directions, respectively. Judging from these results a required accuracy of 0.3 pixels ( $3\sigma$ ) for the inter-telescope registration will be achievable by averaging the image matching data in the same observation unit, if the boresight of each telescope is stable during a maximum observation time of 16 minutes. A hundred data points will be enough for averaging to satisfy a required accuracy of 0.3 pixels in  $3\sigma$ .

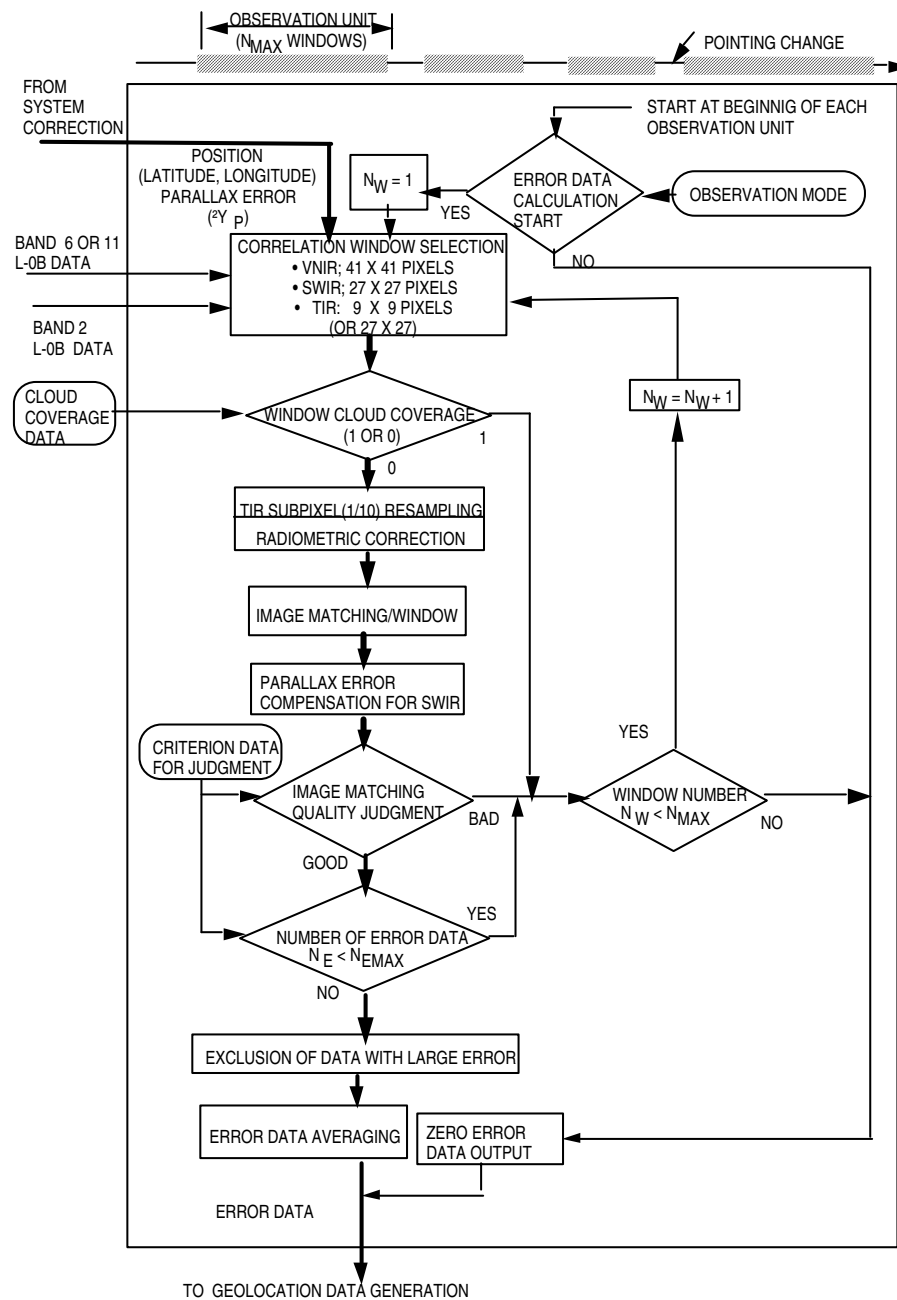


Figure 2-9 Inter-telescope geometric correction flow

## 2.8. Geometric Coefficients Generation

All registration errors which are calculated by the parallax correction and the inter-telescope geometric correction processes are consolidated and changed into latitude/longitude from pixel units in the along- and cross-track directions. The latitude/longitude values at each lattice point, which are calculated by the geometric system correction, are corrected with the consolidated error data.

A set of positions expressed by latitude and longitude are adopted as the geolocation data (the geometric coefficients) for each lattice point.

Other parameters which are necessary for higher level product generations such as Level-3 (the geocoded ortho images) data products and the Level-4 (DEM) data products are calculated in this module, and are appended to the Level-1A products as the header information. Figure 2-10 shows the geometric coefficient generation flow.

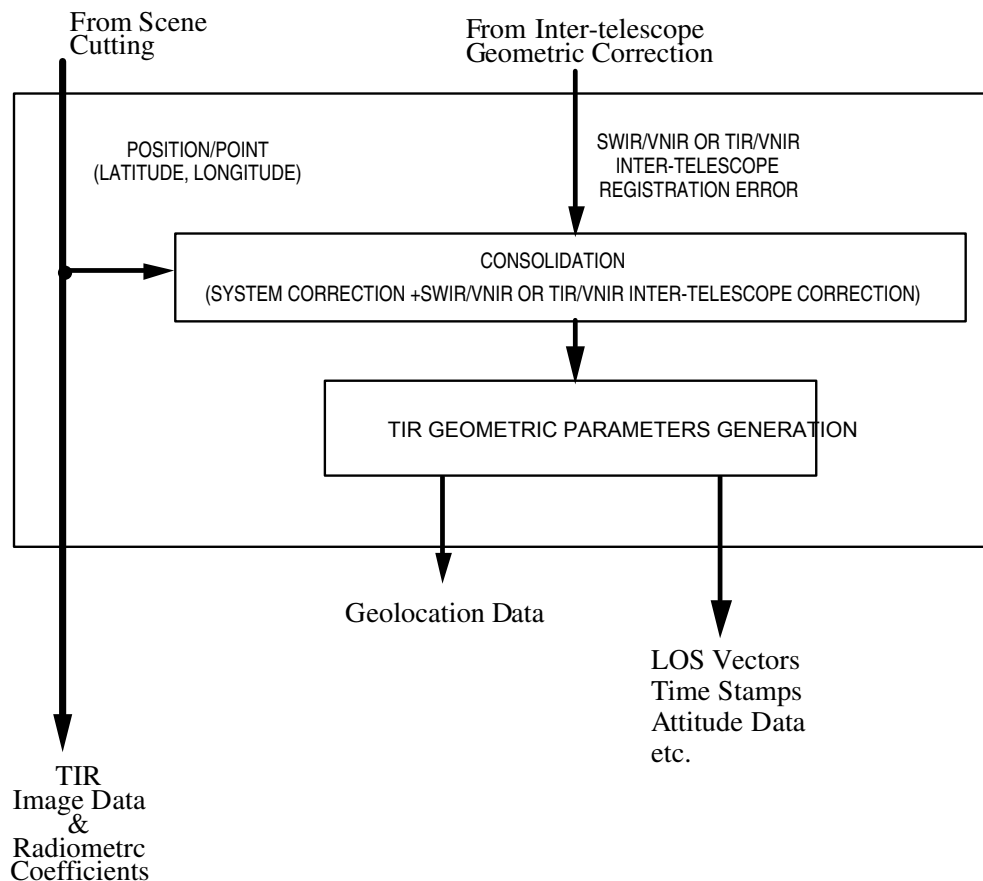


Figure 2-10 Geometric Coefficients Generation

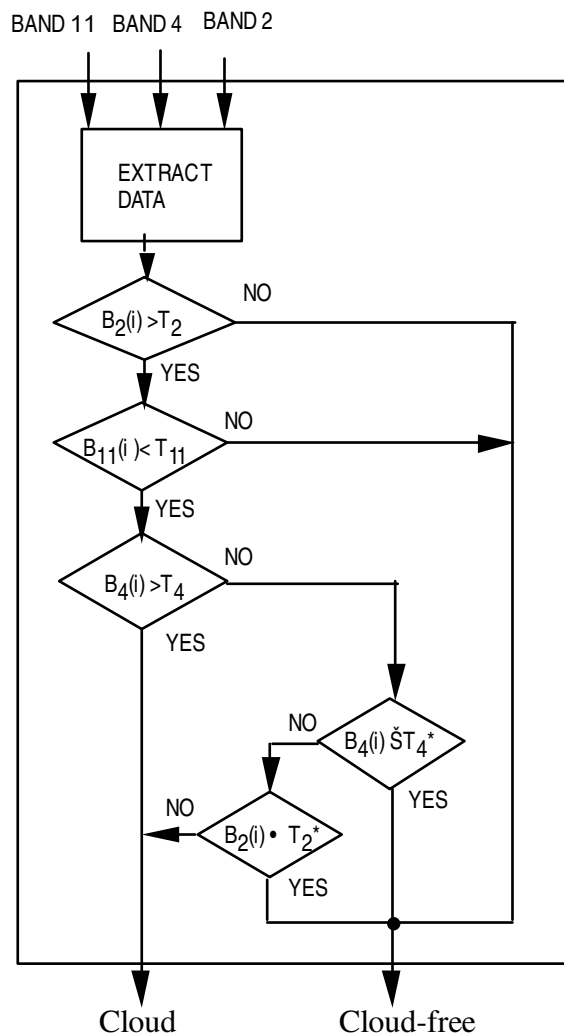
## 2.9. Cloud Coverage Evaluation

The cloud coverage data are used to select the images for image matching in the SWIR parallax correction and the inter-telescope registration, since cloud-free images are essential for image matching. The algorithm is based on the fact that clouds have the highest reflectivity in the visible and the short wave infrared spectral region except for snow and ice, and a low emission in the thermal infrared spectral region because of their lower temperature than targets on the earth.

It will be very important to distinguish clouds from snow and ice on the earth's surface. The discrimination is carried out using knowledge that snow/ice may be brighter in band 2 and darker in band 4 than cloud.

The EOSAT algorithm is employed for cloud coverage calculation. Figure 2-11 (a) shows the algorithm flow. The bands 2, 4 and 11 data are used as representative data for the visible, short wave and thermal infrared spectral regions, respectively. Two threshold values of  $T_2$  and  $T_2^*$ , two threshold values of  $T_4$  and  $T_4^*$  and one threshold value of  $T_{11}$  are introduced for band 2, band 4 and band 11, respectively, as the borders between cloudy and cloud-free targets. According to this algorithm the hatched regions shown in Figure 2-11(b) are judged to be cloudy.

The calculation is carried out for every block whose size is 20 x 20 SWIR pixels. The threshold value depends on the latitude and the season.

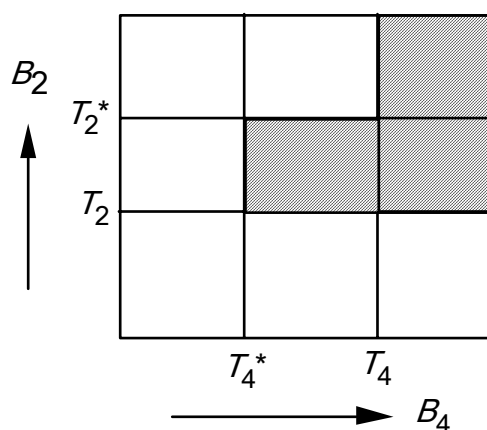


When  $B_{11} \cdot T_{11}$

The image block is judged to be cloud-free regardless of DN values of other bands.

When  $B_{11} < T_{11}$

The hatching areas shown below are judged to be cloudy.



(b) Ranges judged to be cloudy (hatching areas)

$B_2(i)$ : Average DN value of band 2 for block  $i$   
 $B_4(i)$ : Average DN value of band 4 for block  $i$   
 $B_{11}(i)$ : Average DN value of band 11 for block  $i$   
 $T_2, T_2^*$ : Threshold values of band 2  $T_2 \leq T_2^*$   
 $T_4, T_4^*$ : Threshold values of band 4  $T_4 \leq T_4^*$   
 $T_{11}$ : Threshold value of band 11

(a) Cloud assessment algorithm flow

Figure 2-11 Cloud assessment algorithm

## 3. Level-1B Processing Algorithm

### 3.1. Map Projection

Level-1A data product consists of the image data, the radiometric coefficients, the geometric correction coefficients and the auxiliary data. The Level-1B data products will be generated by using these data for the requested map projection and resampling method.

Figure 3-1 shows the pseudo-affine coefficients generation algorithm flow for map projections such as UTM, LCC, SOM, Polar Stereo and Uniform Lat/Long. The coordinate transformation from latitude/longitude to the selected map projection coordinates is followed by the coordinate transformation to Level-1 coordinates according to the pixel size units of each band. The path-oriented coordinates are used rather than the map-oriented coordinates in order to keep the image quality as close to the Level-0 data as possible. The pixel sizes of Level-1 are 15 m for VNIR bands, 30 m for SWIR bands and 90 m for TIR bands on the standard lines for each map projection regardless of real pixel sizes which depends slightly on the spacecraft altitude and pointing angle.

A set of the pseudo-affine transformation coefficients which consists of eight coefficients are generated for each block of Level-1 coordinates by using the relation from the Level-0B to the Level-1 coordinates according to the well-established usual procedure. The size of the block is the same as that of Level-0 coordinates.



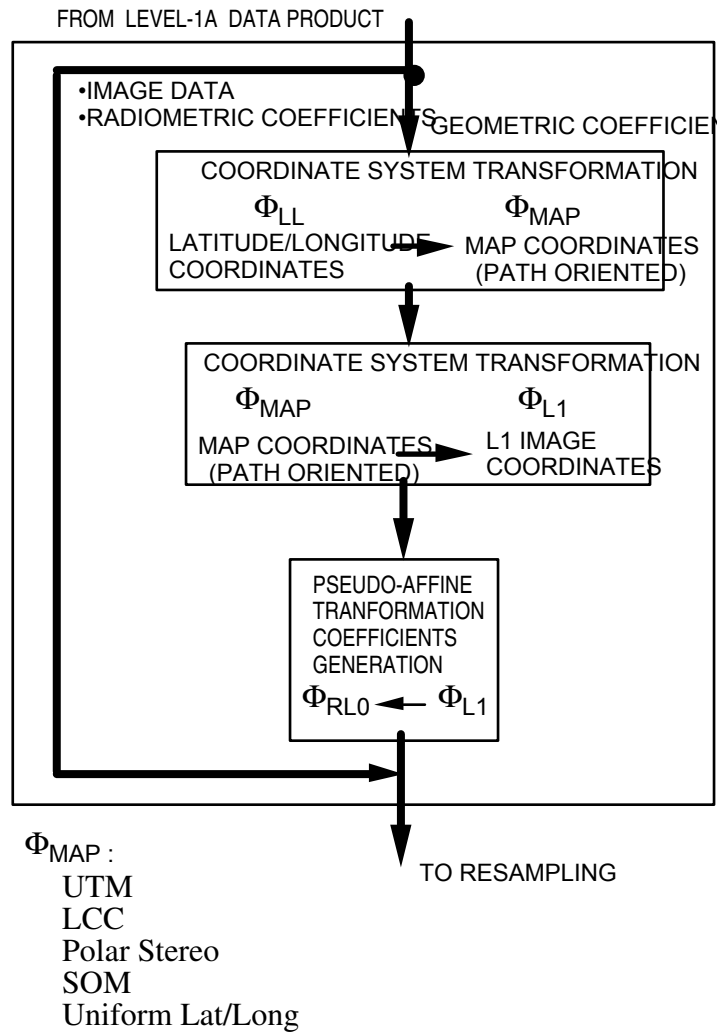


Figure 3-1 Map Projection

## 3.2. Resampling

Figure 3-2 shows the geometric resampling. All pixel addresses in the Level-1 coordinate system are transformed into the realigned (stagger-corrected) Level-0B coordinates using one, four or sixteen pixel addresses, depending on resampling methods selected. Prior to resampling, DN values of bad pixels are evaluated by linear interpolation from the adjacent pixels, followed by the destriping correction.

Resampling is carried out using the radiometric coefficients of detectors. The nearest neighbor (NN), the bi-linear (BL) and the cubic convolution (CC) methods are available types of resampling. Finally the radiance is converted to DN value using the radiance conversion coefficient which are stored in the radiometric data base file.

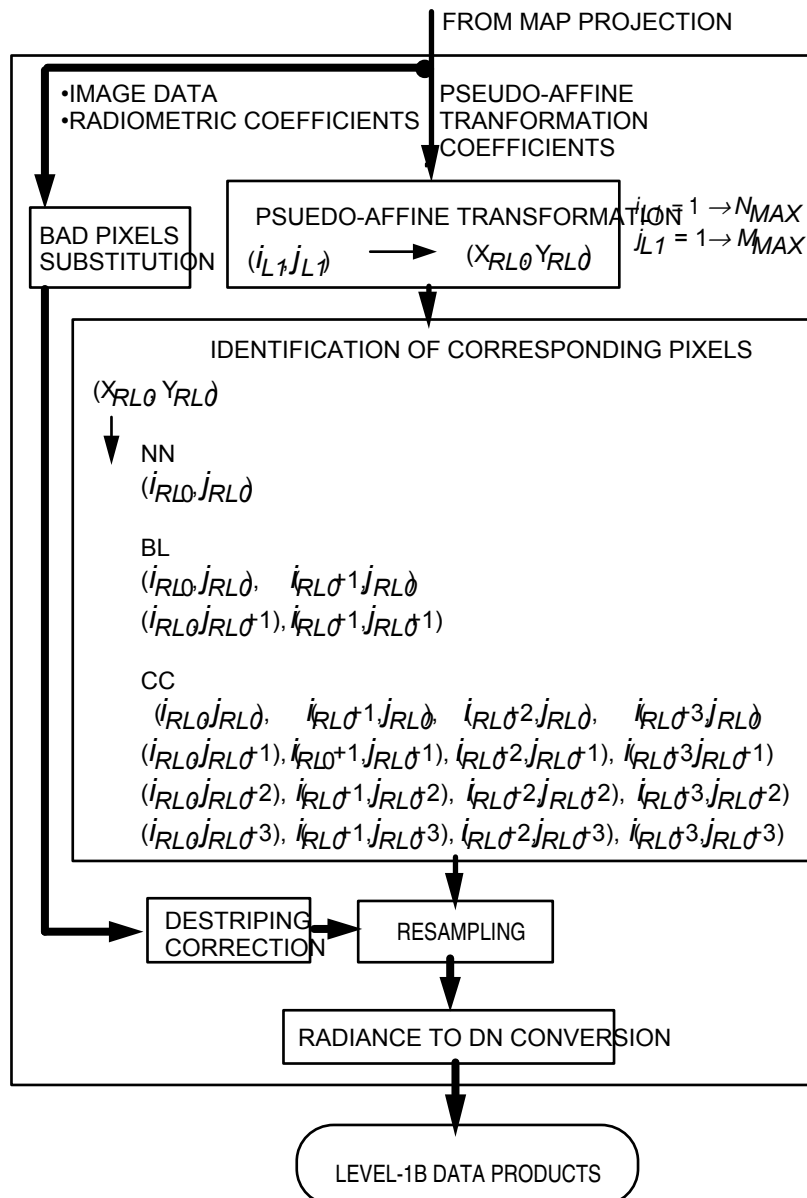


Figure 3-2 Geometric Resampling

## 4. Data Products

### 4.1. Level-1A Data Products

*Scene Definition* : The scene cutting is carried out on Level-0B data according to the predetermined World Reference System (WRS). Each group of data are divided into scenes every 60 km for the along-track direction. One scene size is 63 km to include an overlap of 5 % with neighboring scenes except for band 3B. For band 3B the scene size is 81km, including an additional overlap of 6 km to compensate for the terrain error contribution and a scene rotation for a large cross-track pointing (Figure 4-1).

This scene cutting is necessary for granularizing the Level-1A and Level-1B data products. It does not necessarily mean that the scene position is rigidly predetermined. It is still possible to revert to Level-0B data for a different cutting of the scene.

*Contents Outline*: Figure 4-2 shows the Level-1A data products outline. Level-1A data are formally defined as reconstructed, unprocessed instrument data at full resolution. According to this definition the ASTER Level-1A data consists of the image data, the radiometric coefficients, the geometric coefficients and other auxiliary data, without applying the coefficients to the image data to keep the original data values.

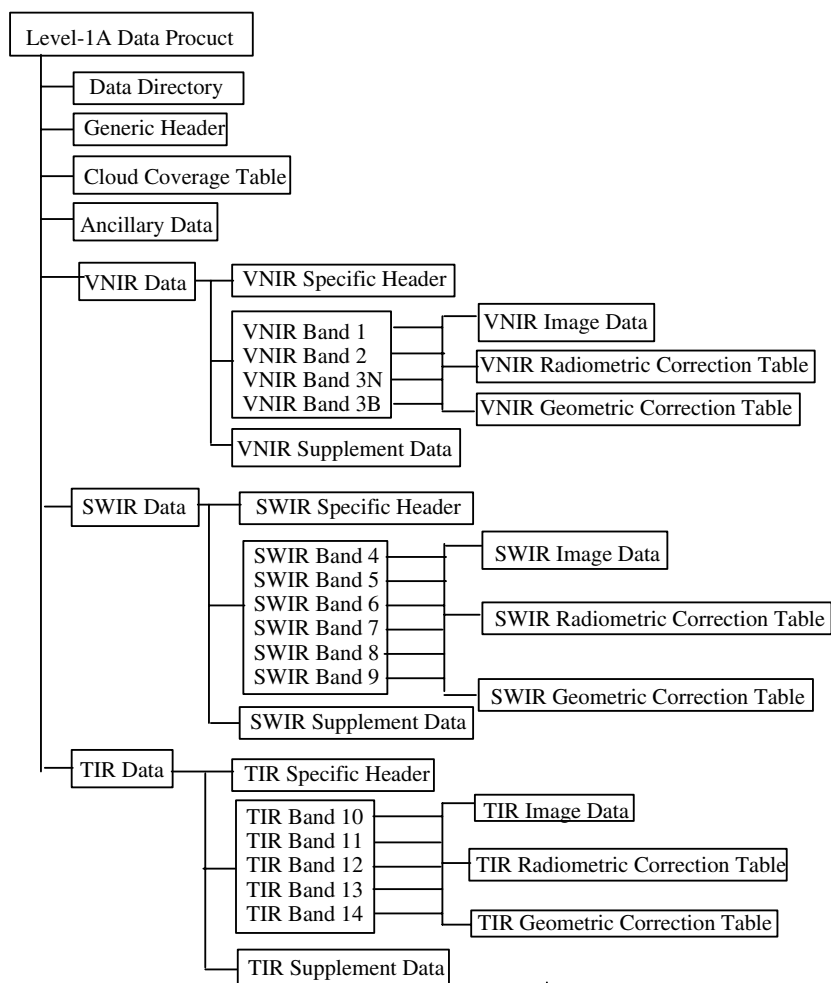


Figure 4-2 Level-1A data product outline

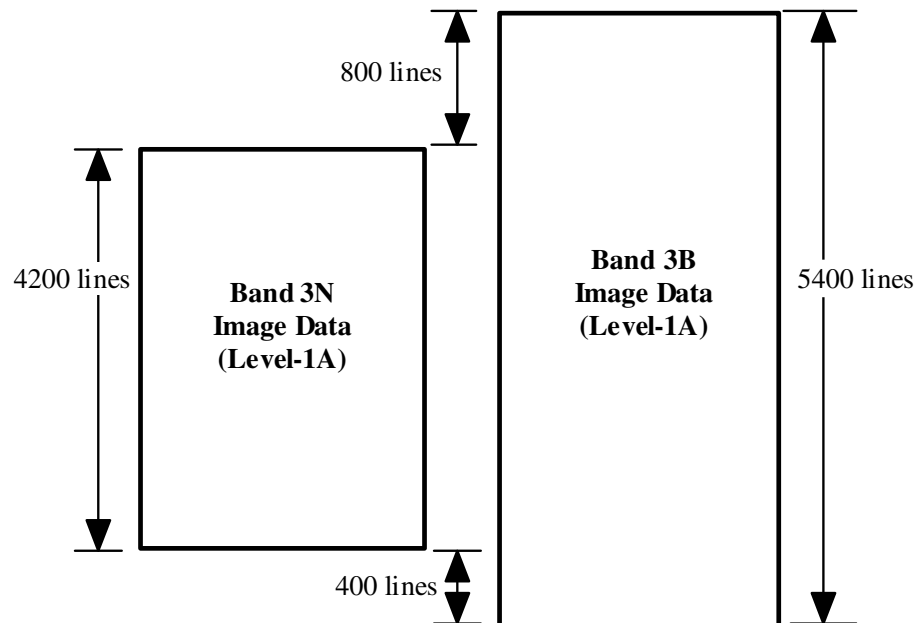


Figure 4-1 Relation between band 3N and band 3B image lines of Level-1A product for an elevation of zero

#### *Data Sizes:*

Table 4-1 shows the Level-1A data product size.

Table 4-1 Level1A Data Product Size

Item	Data size (byte)
Data Directory	8,192
Generic Header	about 4,000
Specific Header	about 10,100
Cloud Coverage Table	1,365
Ancillary Data	about 1,728
Supplement Data	about 1,379,550
VNIR Image Data	74,660,000
SWIR Image Data	25,804,800
TIR Image Data	4,900,000
Radiometric Correction Table	355,656
Geometric Correction Table	4,746,080
Total	about 112 MB

## 4.2. Level-1B Data Products

### *Contents Outline:*

Figure 4-3 shows the outline of Level-1B data.

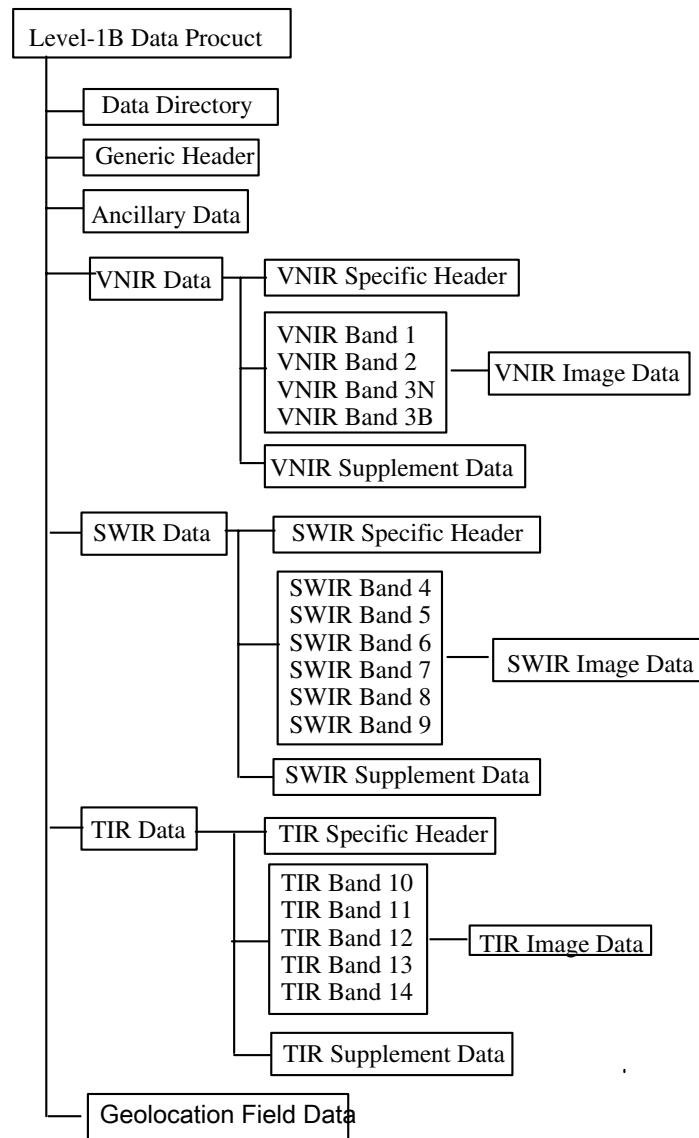


Figure 4-3 Level-1B data product outline

*Data Sizes:*

Table 4-2 shows the Level-1B data product size.

Table 4-2 Level-1B Data Product Size

Item	Data size (byte)
Data Directory	8,192
Generic Header	about 4,000
Specific Header	about 9,100
Ancillary Data	about 1,728
Supplement Data	about 1,379,550
VNIR Image Data	85,656,000
SWIR Image Data	31,794,000
TIR Image Data	5,810,000
Geolocation Data Fiele	TBD
Total	about 125 MB

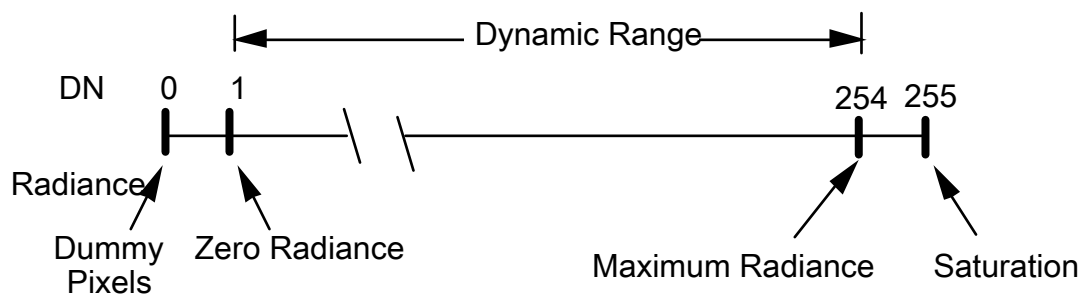
### *Radiometric Parameters:*

Unit conversion coefficients, which is defined as radiance per 1DN, are used to convert from DN to radiance. Radiance (spectral radiance) is expressed in unit of  $W/(m^2 \cdot sr \cdot \mu m)$ . It is the basic policy that the unit conversion coefficient will be kept in the same values throughout mission life.

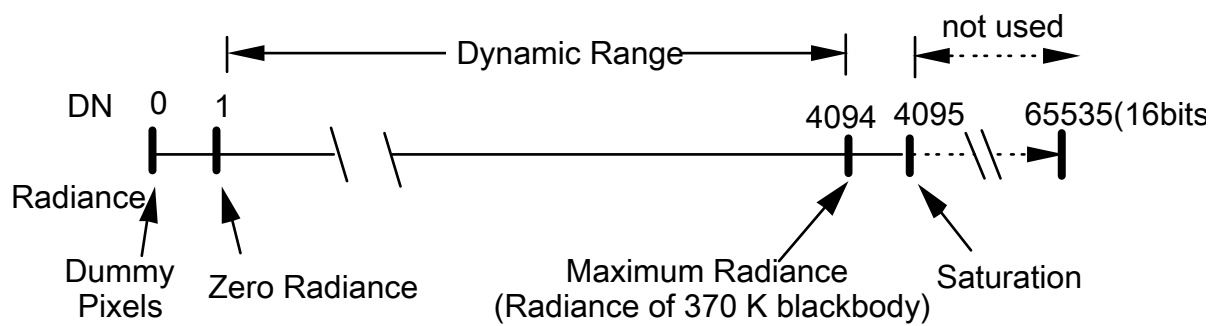
The relation between DN values and radiances is shown below and illustrated in Figure 4-4.

- (i) A DN value of zero is allocated to dummy pixels.
- (ii) A DN value of 1 is allocated to zero radiance.
- (iii) A DN value of 254 is allocated to the maximum radiance for VNIR and SWIR bands.
- (iv) A DN value of 4094 is allocated to the maximum radiance for TIR bands.
- (v) A DN value of 255 is allocated to saturated pixels for VNIR and SWIR bands.
- (vi) A DN value of 4095 is allocated to saturated pixels for TIR bands.





(a) VNIR and SWIR bands



(b) TIR bands

Figure 4-4 Relation between DN values and radiances

The maximum radiances depends on both the spectral bands and the gain settings and are shown in Table 4-3.

Table 4-3 Maximum radiance

Band No.	Maximum radiance (W/(m <sup>2</sup> •sr•μm))			
	High gain	Normal gain	Low gain 1	Low gain 2
1	170.8	427	569	N/A
2	179.0	358	477	
3N	**106.8	218	290	
3B	**106.8	218	290	
4	27.5	55.0	73.3	73.3
5	8.8	17.6	23.4	103.5
6	7.9	15.8	21.0	98.7
7	7.55	15.1	20.1	83.8
8	5.27	10.55	14.06	62.0
9	4.02	8.04	10.72	67.0
10	N/A	28.17*	N/A	N/A
11		27.75*		
12		26.97*		
13		23.30*		
14		21.38*		

Note:

\* Radiance of 370 K blackbody

\*\* Apparent gain is 2.0412 which is slightly different from a nominal high gain value of 2.0.

Maximum radiances for high and low gains are basically defined as those for normal gain divided by nominal gain except for band 3N and 3B for high gain. For band 3N and 3B, the maximum radiance is slightly smaller than the value calculated above which may be saturated because of a large offset..

The unit conversion coefficients can be calculated as follows.

$$\begin{aligned} L_{ni} &= L_{maxi}/253 & (\text{VNIR and SWIR bands}) \\ L_{ni} &= L_{maxi}/4093 & (\text{TIR bands}) \end{aligned} \quad (4-1)$$

where  $L_{ni}$ : the unit conversion coefficient from DN to radiance of band  $i$   
 $L_{maxi}$ : the maximum radiance of band  $i$

Table 4-4 shows the calculated unit conversion coefficients of each band.

Table 4-4 Unit conversion coefficients

Band No.	Coefficient (W/(m <sup>2</sup> •sr•μm)/DN)			
	High gain	Normal gain	Low gain 1	Low gain 2
1	0.676	1.688	2.25	N/A
2	0.708	1.415	1.89	
3N	0.423	0.862	1.15	
3B	0.423	0.862	1.15	
4	0.1087	0.2174	0.290	0.290
5	0.0348	0.0696	0.0925	0.409
6	0.0313	0.0625	0.0830	0.390
7	0.0299	0.0597	0.0795	0.332
8	0.0209	0.0417	0.0556	0.245
9	0.0159	0.0318	0.0424	0.265
10	N/A	6.882 x 10 <sup>-3</sup>	N/A	N/A
11		6.780 x 10 <sup>-3</sup>		
12		6.590 x 10 <sup>-3</sup>		
13		5.693 x 10 <sup>-3</sup>		
14		5.225 x 10 <sup>-3</sup>		

From the relation mentioned above the radiance value can be obtained from DN values as follows.

$$\text{Radiance} = (\text{DN value} - 1) \times \text{Unit conversion coefficient} \quad (4-2)$$

### *Geometric Parameters:*

Parameters related to the geometric properties are map projection, ellipsoid, pixel size and resampling method. Major features of these parameters are as follows.

#### (1) Map Projection

- (i) Map projections are limited to Universal Transverse Mercator (UTM), Lambert Conformal Conic (LCC), Polar Stereographic (PS), Space Oblique Mercator (SOM) and uniform Lat/Long.
- (ii) Map direction is limited to the Path Oriented.
- (iii) For UTM two standard longitude line method is adopted with a reduction rate of 0.9996 to define the cylinder.
- (iv) For LCC two standard latitude lines of 53 and 67 degrees are adopted to define the cone position and angle.
- (v) For PS a standard latitude line of 70 degree is adopted to define the plane position.
- (vi) FOR SOM the nominal orbit path line is used as the position to contact to the projected plane.
- (vii) Default map projection is UTM regardless of latitude.

#### (2) Ellipsoid

The Earth ellipsoid is limited to WGS-84. Therefore, it should be noted that the observation point expressed in latitude and longitude is the intersection of WGS-84 ellipsoid and an extension of the line-of-sight vector. The terrain error is included in the latitude and longitude values caused by a difference between WGS-84 ellipsoid and real earth surface as shown in Figure 4-5.

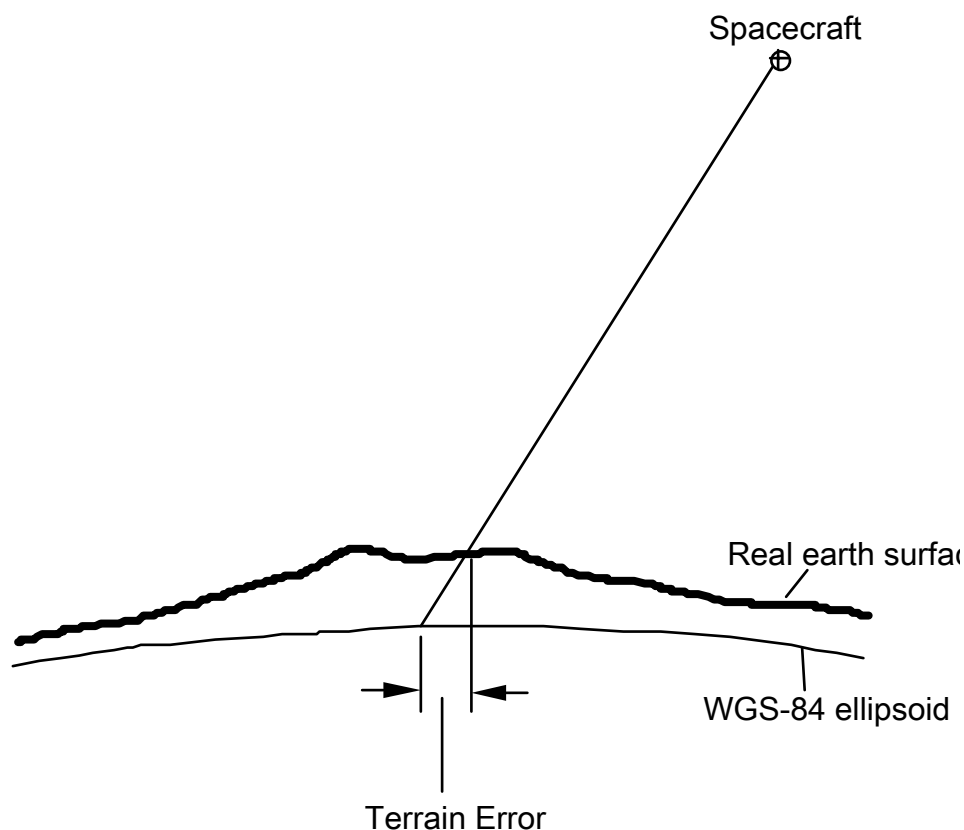


Figure 4-5 Terrain error

### (3) Pixel Size

- (i) It is the basic policy that the map-projected pixel size does not depend on the data acquisition condition such as instrument cross-track pointing and orbit drift but depends only on the earth surface position (latitude/longitude) for each map projection.
- (ii) The nominal pixel sizes are 15 m, 30 m and 90 m for VNIR, SWIR and TIR, respectively.
- (iii) Map-projected pixel sizes meet the nominal sizes along the standard lines.

### (4) Resampling Method

- (i) The nearest neighbor (NN), bi-linear (BL) and cubic convolution (CC) methods are available types of resampling.
- (ii) Default resampling method is the cubic convolution.

### *Bad Pixel Replacement Method:*

#### (1) Definition of bad pixels

- (i) Pixels included in missing packets during the down link.
- (ii) Damaged detectors.
- (iii) All Level-1B pixels generated from bad pixels of Level-1A (Figure 4-6).

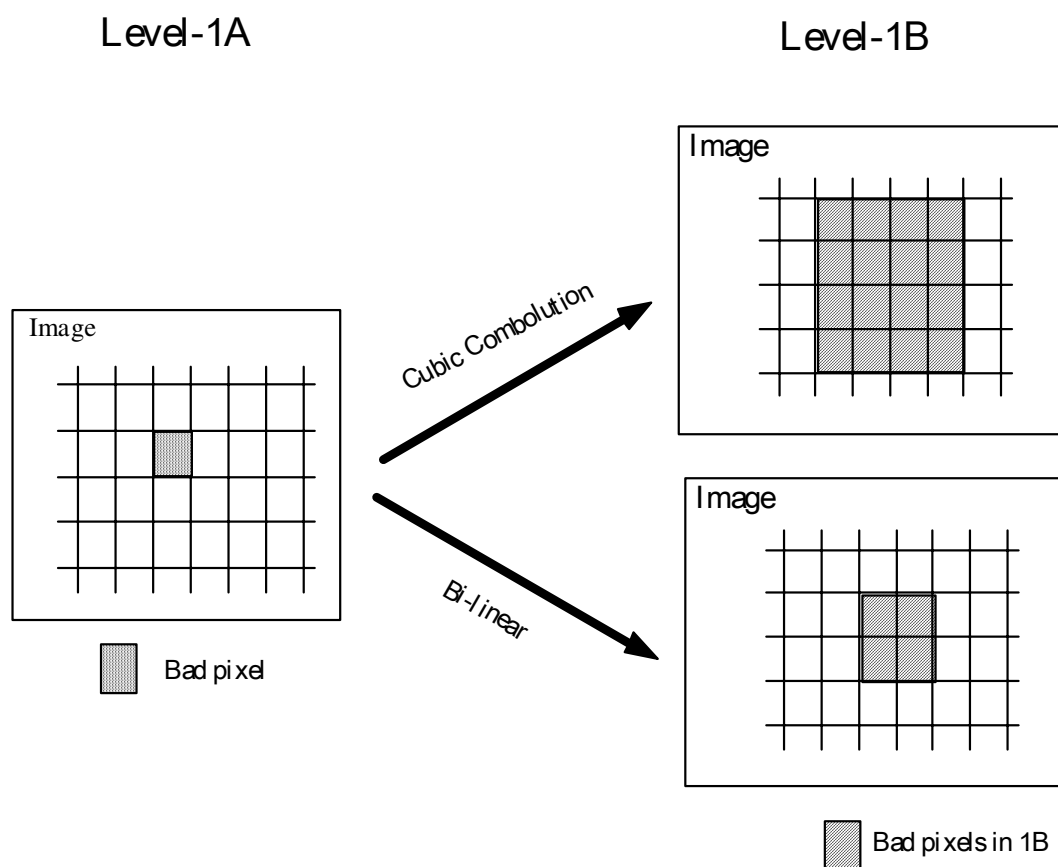


Figure 4-6 Definition of bad pixels in Level-B data products

#### (2) Basic replacement policy

- (i) The bad pixel replacement processing is applied to the Level-1B products generation.
- (ii) The simple linear interpolation method is used to generate the values for the replacement (Figure 4-7).
- (iii) The bad pixels are listed using the first and the last pixels in the cross-track direction in the specific header.

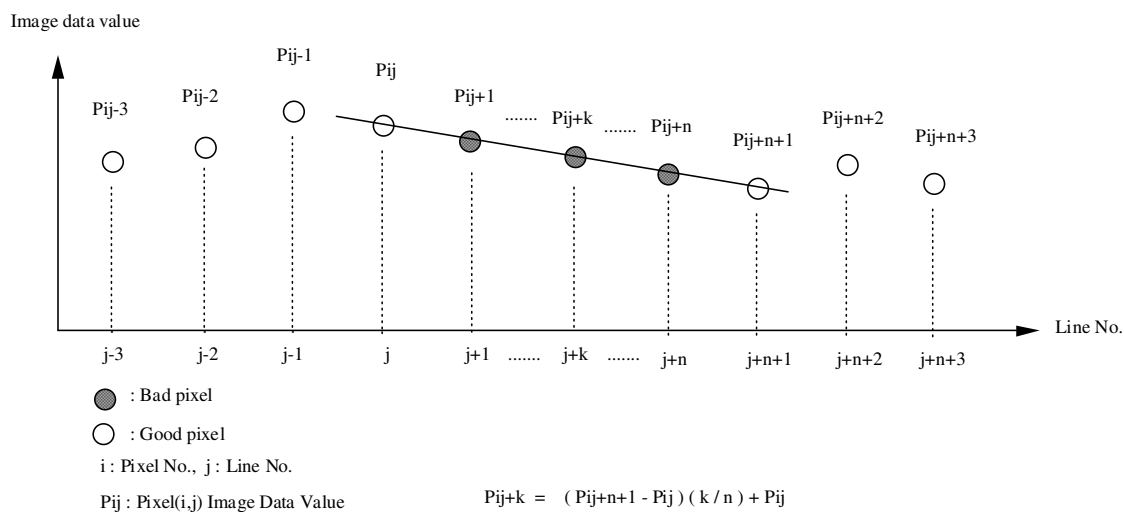


Figure 4-7 Bad pixel replacement in the along track direction

### 4.3. Browse Data Products

*Objective:* Level-1 browse images are provided to overview observed images at reduced resolution, and have image values relative to radiance. Users can know rough situation of cloud/snow coverage and ground surface objects simply by looking the Level-1 browse images. Level-1 browse products are usually used for retrieval of Level-1 products.

*Band Selection and Color Assignment :* One type of color browse images will be generated for each three subsystem (VNIR, SWIR, TIR) in all Level-1 products. Band 1, 2, 3N, Band 4, 5, 9 and Band 10, 12, 14 are assigned to the blue, green and red component in browse images for VNIR, SWIR and TIR subsystems, respectively, as shown in Table 4-5.

*Size of Browse Images:* The resolution of a pixel in Level-1 images of all subsystem are reduced into about 309 m/pixel using average sampling. The sampled image is put in a frame having the size of 224 pixels by 208 lines. This frame size of a browse image corresponds with about 5.1 cm by 4.7 cm on the display monitor when a 17 inch monitor with XGA mode(1024 x 768 dots) is used.

*Orientation of Browse Images:* Geometric correction including map projection is not performed for browse image. The browse images are always arranged to direct the north to upper half part regardless of the spacecraft flight direction (descending or ascending) as shown in Figure 4-7. And skew distortion caused by the earth rotation is also corrected. Figure 4-8 shows orientation of browse image and Level-1A images

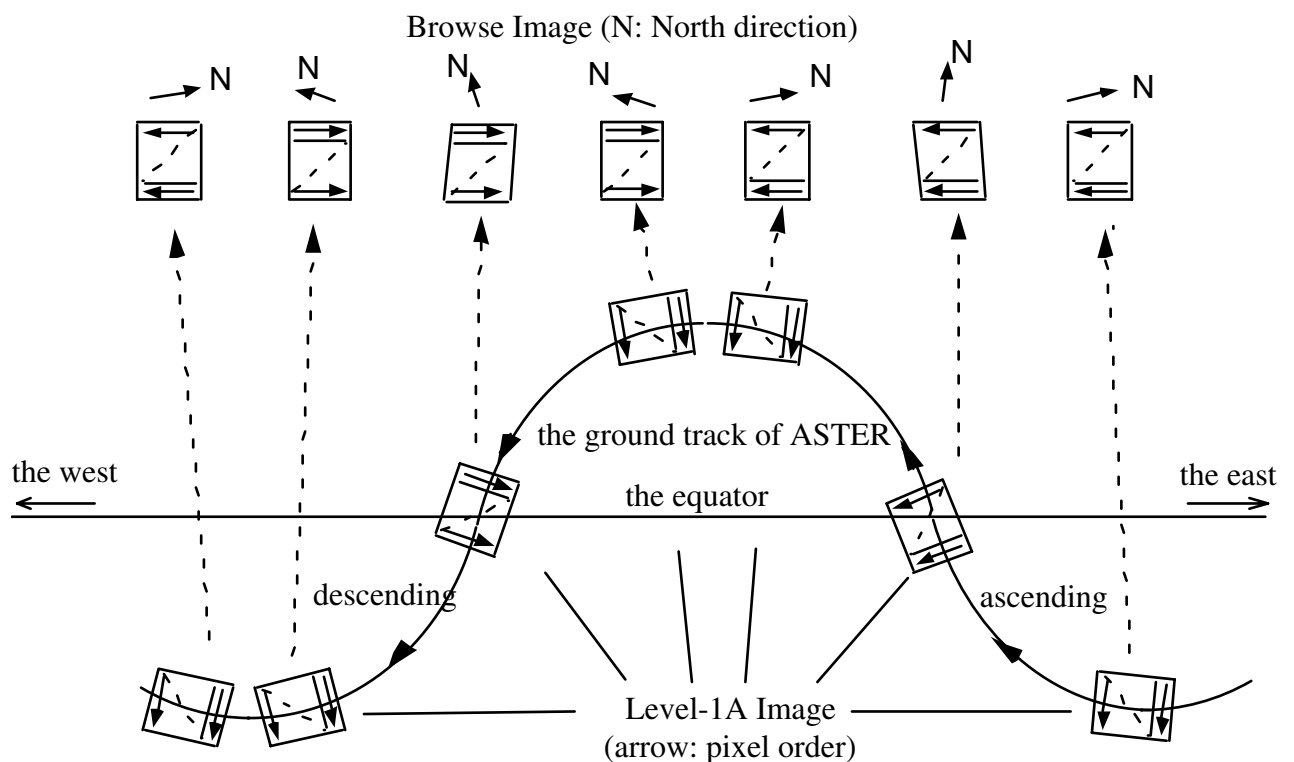


Figure 4-8 Orientation of browse images



*Radiometric Correction:* Level-1A data values are radiometrically corrected by the same manner of radiometric correction conducted in Level-1B. Then, the radiance is rescaled into 8 bits range using linear density stretch. Finally, the 8 bits color image is compressed using JPEG lossy method. The data volume decreases in about 1/15 in usually.

Table 4-5 Specifications of Level-1 Browse Images.

Target Data				
target products	all Level 1A products			
target subsystems	one browse image per VNIR, SWIR and TIR respectively 3 bands per a subsystem			
target bands	default bands combination: VNIR=1,2,3N, SWIR=4,5,9, TIR=10,12,14			
Radiometric Correction	the same radiometric correction as Level 1B processing			
Contrast Conversion				
conversion method	linear contrast conversion			
conversion coeff.	decided based on percentile method and simple observation model (named as median method)			
Image Type of Browse	color image			
Color Assignment	Default assignment.	VNIR	SWIR	TIR
	B	1	4	10
	G	2	5	12
	R	3N	9	14
Geometric Correction				
map projection	no apply			
skew correction	done			
bands registration	Intra/inter-bands registration			
others	Stagger alignment and sampling delay are considered.			
Image Size of Browse				
frame size	224 pixels * 208 lines (5.1cm * 4.7cm on a display monitor: when 17" monitor and 1024dots/window)			
effective size	about 199 pixels * 204 lines for VNIR and SWIR about 204 pixels * 204 lines for TIR (4.5-4.6cm * 4.6cm on user's display monitors :when 17"monitor and 1024 dots/window)			
Sampling				
sampling rate	about 1/20.59(VNIR), 1/10.29(SWIR), 1/3.43(TIR)			
sampling interval	about 309m on the ground			
sampling method	average sampling			
Data Compression				
compression method	JPEG			
Q-value	50			
compression ratio	about 1/15 (estimation)			

#### **4.4. HDF**

TBD

## 5. Data Quality Information

Table 5-1 summarizes the required and the estimated values of the radiometric and the geometric accuracies in preflight phase

Table 5-1 Summary of accuracy

Item	Requirement	Estimation			
Absolute radiometric accuracy (1 $\sigma$ ) (VNIR & SWIR)	4 %	<u>Preflight</u>			
		2.0 % (VNIR)			
		2.9 % (SWIR)			
		<u>On-board</u>			
		2.4 % (VNIR)			
Absolute temperature accuracy (1 $\sigma$ ) (TIR)	3 K (200 - 240 K) 2 K (240 - 270 K) 1 K (270 - 340 K) 2 K (340 - 370 K)	<u>Total</u>			
		Target temperature	200K	300K	370K
		Band 10	2.4K	0.3K	0.8K
		Band 11	1.5K	0.3K	0.5K
		Band 12	1.5K	0.3K	0.5K
		Band 13	0.4K	0.3K	0.1K
		Band 14	0.3K	0.3K	0.1K
Geolocation accuracy (3 $\sigma$ ) (nadir direction)	431 m (AT) 437 m (CT)	47 m (AT) 54 m (CT)			
Band-to-band registration (3 $\sigma$ ) (intra-telescope)	0.2 pixels	<0.2 pixels (VNIR, TIR) about 0.3 pixels (SWIR)			
Band-to-band registration between reference bands (3 $\sigma$ ) (inter-telescope)	0.3 pixels of coarser band	<u>VNIR-SWIR</u>			
		0.051 SWIR pixels (AT)			
		0.054 SWIR pixels (CT)			
		<u>VNIR-TIR</u>			
		0.044 TIR pixels (AT) 0.050 TIR pixels (CT)			

Note AT: along-track direction, CT: cross-track direction

## 6. Calibration/Validation Activities

*Outline:* The Level-1 data processing structure is composed of two parts, processing modules and on-line database files. Each module function is carried out using contents in the Level-0 data and processing parameters accommodated in the on-line database files. The on-line database files accommodate processing parameters which may need revision throughout operation period to keep data quality. The radiometric and geometric database files are major parts of the on-line database files from the data quality point of view.

Preparation and update of the database files are carried out by ASTER Science Team through the calibration/ validation activities which include the on-board calibration and the acquired image analysis. The calibration/validation activity flow is shown in Figure 6-1. On-board calibration is carried out every 17 days in most frequent case during the normal operation period. The image analysis for the vicarious calibration, the band-to-band registration and GCP matching will be carried out less frequently.

The version information is accommodated in the generic header of the Level-1 data product

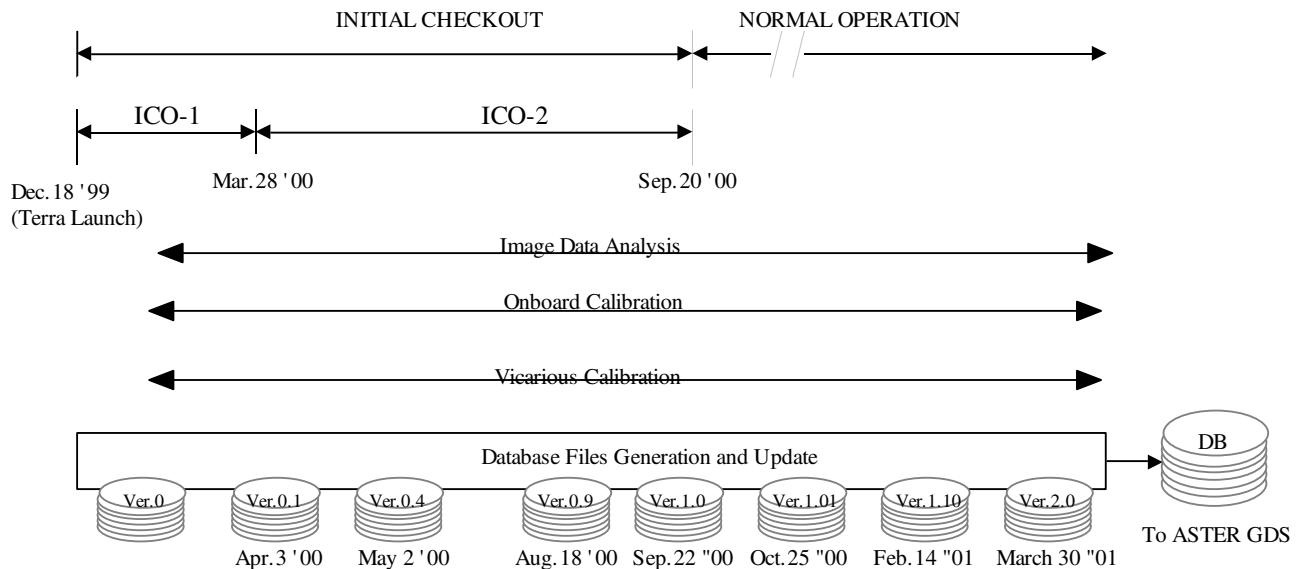


Figure 6-1 Database calibration/validation activity flow

**On-line Data Base Update:** Figure 6-2 shows the relation between the major Level-1 processing modules and the on-line data base files. The on-line data base files will be updated based on the validation results.

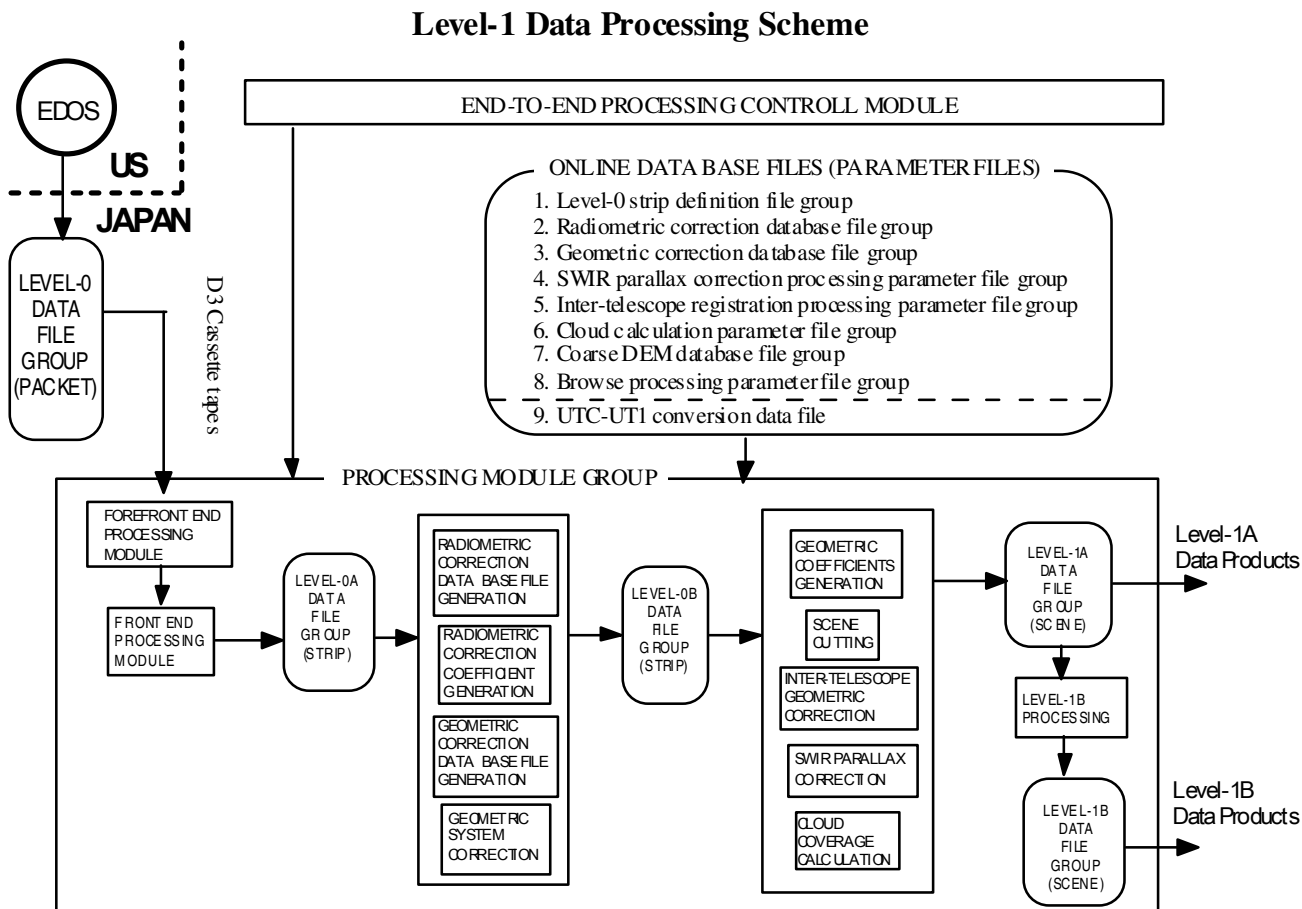


Figure 6-2 Relation between the major Level-1 processing modules and the on-line data base files

### ***Definition of Database File:***

- Version 0 : Prepared from ground measurements during preflight phase.
- Version 0.4 : Updated using very initial images during the early initial checkout period for the Level-1 processing system tests.
- Version 1 : Updated using various calibration and validation data during the initial checkout period and used as the first version for the Level-1 data product generation in the normal operation period.
- Version N : Updated using the calibration and validation data in the normal operation period.

### ***Format of Geometric DB version::***

Description: This provides the version information of the geometric correction data such as Line-Of-Sight vectors and scan constants, which are used to generate Geometric Correction Tables. This includes the following;

- Version No.
- Issuance Date
- Comments

Format:: ( Version , IssuanceDate , “Comments” )

Version = string(10)

IssuanceDate = YYYY-MM-DD

Comments = double quoted string

Example: GeometricDBversion = ( 1.01 , 1999-12-31 , “ xxxxxxxx “

### ***Format of Radiometric DB version:***

Description: This parameter provides the version information of the radiometric correction coefficients, which are used to generate Radiometric Correction Tables.

- Version No.
- Issuance Date
- Comments
- Version
- Issuance Date
- Comments

Format:: ( Version , IssuanceDate , “Comments” )

Version = string(10)

IssuanceDate = YYYY-MM-DD

Comments = double quoted string

Example: RadiometricDBversion = ( 1.01 , 2000-01-01 , “ xxxxxxxx “ )

**Status of Data Quality:** Table 6-1 summarizes the quality of the Level-1 products at June 2001. The version 1.0 and newer version products are publicly released. The following home pages will accommodate the latest information.

ASTER Science Home Page: <http://www.science.aster.ersdac.or.jp/>

ASTER GDS Home Page: <http://www.gds.aster.ersdac.or.jp/>

Table 6-1 Geometric Data Quality (May 1, 2001)

Item		Geometric Online DB Ver.1.02	Geometric Online DB Ver.2.0
Intra-telescope Registration	VNIR	< 0.2 pixels	< 0.1 pixels
	SWIR	< 0.2 pixels	< 0.1 pixels
	TIR	< 0.2 pixels	< 0.1 pixels
Inter-telescope Registration	SWIR/VNIR	< 0.2 pixels	< 0.2 pixels
	TIR/VNIR	< 0.5 pixels	< 0.2 pixels
Stereo Pair System Error	3B/3N	< 65 m	< 10 m
Pixel Geolocation Knowledge*	Relative	< 15 m	< 15 m
	Absolute	< 50 m	< 50 m